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SECURITY CLASSIFICATION OF THIS PAGE (When Data Entered)		REPORT DOCUMENTATION PAGE		READ INSTRUCTIONS REFERRING COMPLETING FORM	
1. REPORT NUMBER		2. GOVT ACCESSION NO.		3. REPORT TYPE AND DATES COVERED	
4. TITLE AND SUBTITLE		5. AUTHOR(s)		6. PERFORMING ORG. REPORT NUMBER	
7. DEVELOPMENT OF A SIX DEGREE OF FREEDOM MOTION SIMULATION MODEL FOR USE IN SUBMARINE DESIGN ANALYSIS.		8. CONTRACT OR GRANT NUMBER(s)		9. PROGRAM ELEMENT, PROJECT, TASK WORK UNIT NUMBERS	
10. PERFORMING ORGANIZATION NAME AND ADDRESS		11. REPORT DATE		12. NUMBER OF PAGES	
MASS. INST. OF TECHNOLOGY		MAY 78		87	
13. CONTROLLING OFFICE NAME AND ADDRESS		14. MONITORING AGENCY NAME & ADDRESS (if different from Controlling Office)		15. SECURITY CLASS (of this report)	
CODE: 031		NAVAL POSTGRADUATE SCHOOL MONTEREY, CALIFORNIA 93940		UNCLASS	
16. DISTRIBUTION STATEMENT (of this Report)		17. DISTRIBUTION STATEMENT (of the abstract entered in Block 20, if different from Report)		18. SUPPLEMENTARY NOTES	
APPROVED FOR PUBLIC RELEASE; DISTRIBUTION UNLIMITED		APPROVED FOR PUBLIC RELEASE; DISTRIBUTION UNLIMITED		DDC DDC DEC 19 1978 A	
19. KEY WORDS (Continue on reverse side if necessary and identify by block number)		20. ABSTRACT (Continue on reverse side if necessary and identify by block number)		21. SUPPLEMENTARY NOTES	
MOTION SIMULATION; SUBMARINE DESIGN; SIX DEGREE OF FREEDOM		SEE REVERSE.		251450 Gm	
22. SECURITY CLASS (of this report)		23. SECURITY CLASS (of this report)		24. SECURITY CLASS (of this report)	
UNCLASS		UNCLASS		UNCLASS	

DD FORM 1 JAN 73 1473
(Page 1)

EDITION OF 1 NOV 65 IS OBSOLETE
S/N 6107-510-6001

UNCLASS
SECURITY CLASSIFICATION OF THIS PAGE (When Data Entered)

DEVELOPMENT OF A SIX DEGREE OF FREEDOM
MOTION SIMULATION MODEL FOR USE IN
SUBMARINE DESIGN ANALYSIS

by

John Thomas Hammond

Submitted to the Department of Ocean Engineering on
May 12, 1978 in partial fulfillment of the requirements for
the Degrees of Ocean Engineer and Master of Science in Naval
Architecture and Marine Engineering.

ABSTRACT

A computer mathematical model that will simulate the six degree of freedom motion of a submarine has been developed. Hydrodynamic coefficients obtained from the testing of physical models enable the user to accurately simulate specific submarine designs. The simulation model can be used as a design tool to study and predict submarine dynamic response. Complete three dimensional motion is allowed for the submarine while constraints in powering, rudder deflection, dive plane angle, and response time can be selected by the user. Outputs include a chronological history of all linear and angular velocities and accelerations, the instantaneous angle of deflection of each control surface, and the trajectory of the submarine center of gravity.

Thesis Supervisor: Martin A. Aukowitz

Title: Professor of Ocean Engineering

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MOTION SIMULATION MODEL FOR USE IN
SUBMARINE DESIGN ANALYSIS

by

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Lieutenant, United States Navy
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(1971)

Submitted in partial fulfillment
of the requirements for the
degrees of

Ocean Engineer
and
Master of Science
in Naval Architecture
and Marine Engineering

at the

Massachusetts Institute of Technology
May, 1978

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Signature of Author
Department of Ocean Engineering May 12, 1978

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Chairman, Department Committee

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Title: Professor of Ocean Engineering

ACKNOWLEDGEMENTS

This thesis was prepared under the auspices of the Department of Ocean Engineering and Professor Martin A. Abkowitz. My sincerest thanks to Professor Abkowitz for his expert guidance in developing the simulation model used in this thesis. I would also like to express special thanks to my wife Robin for her loving support during the months of thesis preparation and for her endless hours of expert typing.

J.T.H.

MAY 1978
CAMBRIDGE, MASSACHUSETTS

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CHAPTER I - INTRODUCTION

I.1 BACKGROUND

During the several years spent in designing a submarine, a great deal of effort is expended trying to improve the design so that the best possible vessel is eventually constructed. During the design phases the designer will frequently examine problems that were found in older submarines and try to determine the cause of each problem so that they can be eliminated in the new design. At several stages during the design, computer models are used to assist in such things as structural design, equilibrium calculations, and internal arrangement. As the design progresses, physical models of the proposed vessel are built and tested in a towing tank to measure the hydrodynamic coefficients. Finally, when it is possible to deal with the vessel as a whole, a dynamic analysis is conducted to gauge how the submarine will perform in an underwater environment when all six degrees of freedom are available.

While it is possible to gather dynamic information, such as the hydrodynamic coefficients, from physical model tests, the models are necessarily too restricted in their motion to permit a complete

analysis. It is necessary to construct a mathematical model of the submarine to simulate the submarine's motion and thereby obtain enough data for a complete analysis. The simulation model will use the hydrodynamic coefficients obtained from the physical model as a basis for the simulation. A properly constructed model will permit the designer to simulate any conceivable maneuver and gauge the submarine's response. The simulation model then becomes a tool to assist in improving the design and achieving the best possible vessel. After the submarine is designed, the model can continue to be useful by assisting in the evaluation of operating procedures. The model can be placed in any maneuvering situation without hazard to crew or vessel. This is especially useful in evaluating casualty situations. The model can also be used to estimate the effect of proposed design changes to the submarine. For instance, the model can show the effect of changing the maximum deflection of a control surface or its rate of operation.

All of the few six degree of freedom submarine simulation models in existence are too expensive for daily use in the design office. Generally, the models are complicated to use and sometimes difficult to obtain. These problems have created the need for a program that is both inexpensive enough to permit daily use in the design office and simple enough so that anyone can use it. This need has formed the motivation for this thesis.

1.2 MODEL DEVELOPMENT

A mathematical model must use Newton's Law of Motion and the

differential equations resulting from a dynamic analysis of the submarine. A submarine has movable appendages, so these must also be accounted for in the model. Newton's Law of Motion can be expressed as:

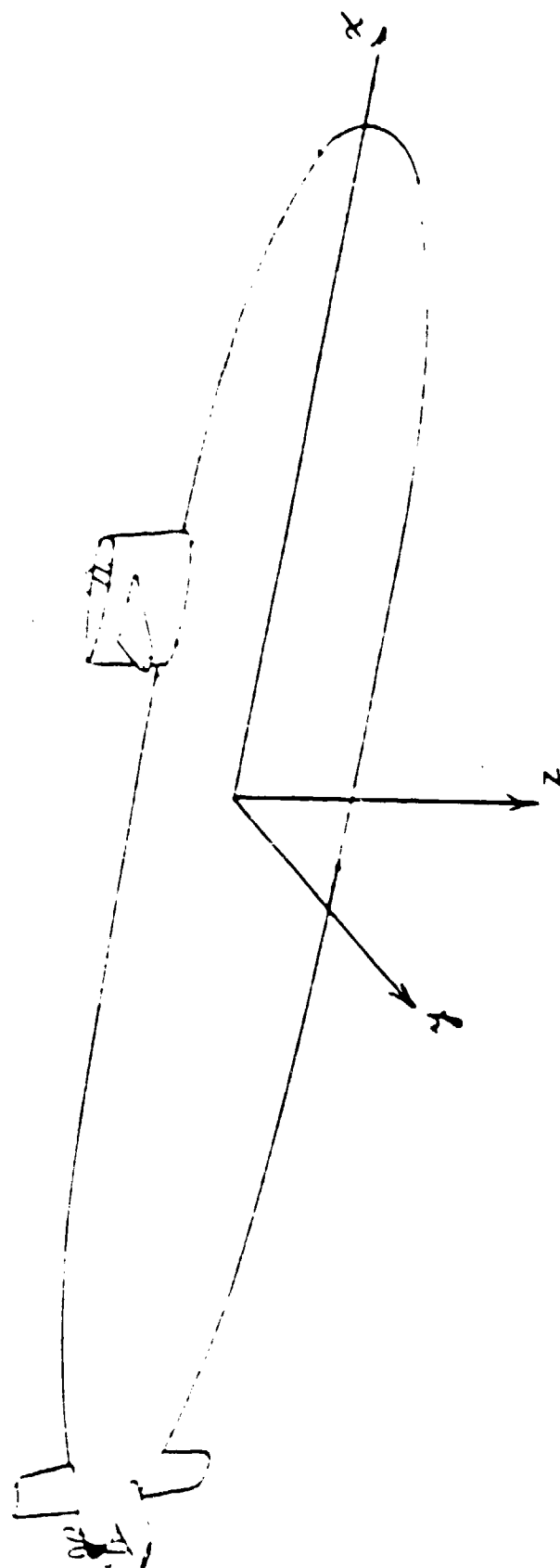
$$(1) \quad \text{Force} = \frac{d}{dt} (\text{momentum})$$

$$(2) \quad \text{Moment} = \frac{d}{dt} (\text{angular momentum})$$

If a submarine with a coordinate system such as shown in Figure 1 is used, then equation (1) can be applied along each of the three axes. Similarly, equation (2) can be applied around each axis. This gives a total of six equations to represent the six degrees of freedom of the submarine. These six equations of motion are well-known [1] so they are not developed in detail here. They are, however, included for reference in Appendix A.

While the equations of motion for the submarine form the heart of the model, they are insufficient by themselves. On an actual submarine, the officer of the deck orders the rudder or dive planes moved in order to maneuver the ship. The routines will normally sense where the ship currently is and where it is supposed to be and then move the appendages in the appropriate direction just as the deck officer would do. The six equations of motion together with the appendage control subroutines constitute the vital components of the simulation model. A main program is necessary to coordinate the input, output, and to control the action of the subroutines.

The components of the model are discussed in detail in Chapter II of this thesis. The model is written as a computer program whose features are discussed in Chapter III. The final chapter will



Local Submarine Coordinate System

Figure 1

discuss the model tests and their results. A listing of the program will be provided in Appendix B.

CHAPTER II - THE SIMULATION MODEL

II.1 GENERAL CAPABILITIES

The simulation model developed in this thesis will provide trajectory information for routine submarine maneuvers such as turning or changing depth. The trajectory, which is referenced to a fixed coordinate system, is computed as a function of time. As the trajectory of the model is developed, the velocities and accelerations are calculated and stored for output to the user. Both angular and linear velocities and accelerations are provided. The angle of deflection of each control surface appendage is computed for each step of the entire trajectory or maneuver.

The rate of deflection of each control surface and its angle of maximum deflection are specified by the user. Any of the control surfaces can be "jammed" by specifying a particular angle of deflection. The model can be initially placed on any course and depth and then ordered to come to any new course and depth. The user can select the initial speed and the appropriate thrust coefficients to cause a change in speed.

II.2 COORDINATE SYSTEMS

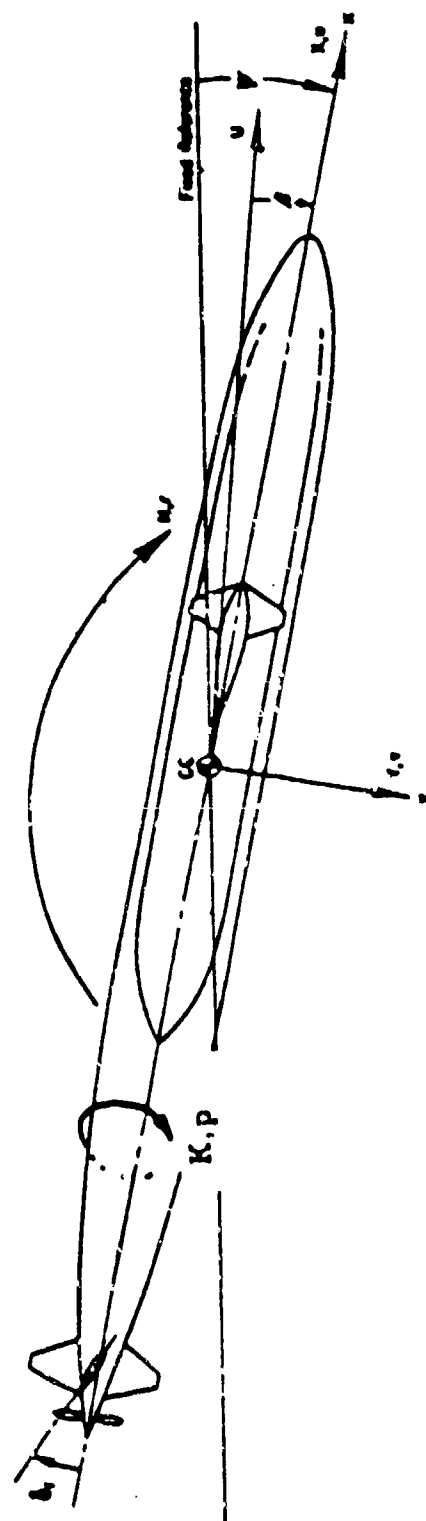
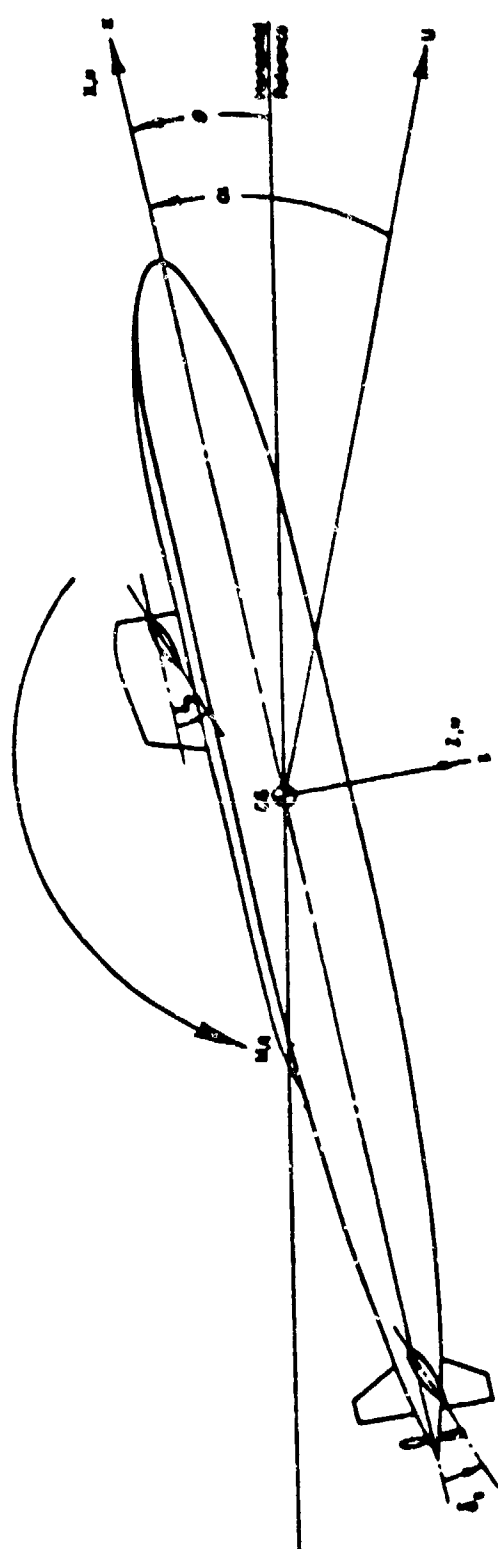
The coordinate system plan used in the simulation model is based on [1]. The model uses two orthogonal coordinate systems; one remaining fixed at the water surface while the other travels with the submarine to act as a local reference system. The fixed system, designated x_0, y_0, z_0 , is related to the moving system, designated x, y, z , through the angles ϕ, θ, ψ as illustrated in Figure 2. Quantities measured in the moving system can be referenced to the fixed system by using the transformation [2] given in Appendix A.

The origin of the moving system is at the center of gravity of the submarine. This has the advantage that only the principal moments of inertia, I_x, I_y, I_z , are non-zero (i.e.: $I_{xy}=I_{yz}=I_{xz}=0$). It has the additional advantage that the equations of motion in [1] also use this reference point for the x, y, z system. It would have been possible to use the centerline of the submarine for the origin. This would have the advantage of making better use of the symmetry of the vessel, however it would have the disadvantage of having to correct both the equations of motion and some of the hydrodynamic coefficients for the new origin.

Appendage movements are measured with respect to the moving coordinate system. All velocities and accelerations are measured along the axes of the moving system. This is also shown in Figure 2.

II.3 EQUATIONS OF MOTION

The general nature of the six degree of freedom model requires the use of the six equations in their non-linear form. A large



Relationship of Axes, Angles, Velocities, Forces, & Moments [1]

Figure 2

number of hydrodynamic coefficients are necessary for this model. It would be desirable to derive each coefficient from existing hydrodynamic theory. Some coefficients have been obtained for bodies of revolution or other mathematically amenable shapes; however, when appendages such as control surfaces, fairwaters, and propellers are taken into account, the accuracy of the theoretical values is brought into question. For this reason it is standard practice to obtain the numerical value of the coefficients by experimental means. This usually entails the towing and measurement of physical models.

Once the hydrodynamic coefficients are known, the equations can be solved for the accelerations. The six equations of motion must be written such that the highest order derivatives, namely \dot{u} , \dot{v} , \dot{w} , \dot{p} , \dot{q} , \dot{r} and their coefficients, appear on the left hand side of the equation. This gives each equation a form like:

$$a_{1,j} \dot{u} + a_{1,j+1} \dot{v} + a_{1,j+2} \dot{w} + a_{1,j+3} \dot{p} + a_{1,j+4} \dot{q} + a_{1,j+5} \dot{r} \\ = f_j (u, v, w, p, q, r, \phi, \theta, \psi, \delta_r, \delta_b, \delta_s, l, m, \dots)$$

The left hand side is placed in a matrix format. The six equations are then represented as:

$$\begin{bmatrix} a_{1,j} \end{bmatrix} \begin{bmatrix} \dot{u} \\ \dot{v} \\ \dot{w} \\ \dot{p} \\ \dot{q} \\ \dot{r} \end{bmatrix} = \begin{bmatrix} f_1 \\ f_2 \\ f_3 \\ f_4 \\ f_5 \\ f_6 \end{bmatrix}$$

The method of solution is an iterative technique in which initial values are used in the right hand side functions and the six accelerations are solved for on the left side. The accelerations are then used to update the right side functions. A small time increment is made and the accelerations are again solved by using the latest value for the right side functions.

With each iteration the accelerations are used in a Taylor series expansion to calculate the velocities. The calculations have the form:

$$\begin{Bmatrix} u \\ v \\ w \\ p \\ q \\ r \end{Bmatrix} (t + \Delta t) = \begin{Bmatrix} u \\ v \\ w \\ p \\ q \\ r \end{Bmatrix} (t) + \begin{Bmatrix} \dot{u} \\ \dot{v} \\ \dot{w} \\ \dot{p} \\ \dot{q} \\ \dot{r} \end{Bmatrix} \cdot (\Delta t)$$

The pitch, roll, and yaw angles are calculated in a similar manner using the angular velocity and acceleration.

$$\begin{Bmatrix} \phi \\ \theta \\ \psi \end{Bmatrix} (t + \Delta t) = \begin{Bmatrix} \phi \\ \theta \\ \psi \end{Bmatrix} (t) + \begin{Bmatrix} p \\ q \\ r \end{Bmatrix} \cdot (\Delta t) + \begin{Bmatrix} \dot{p} \\ \dot{q} \\ \dot{r} \end{Bmatrix} \cdot \left(\frac{\Delta t^2}{2} \right)$$

In order to plot the trajectory of the submarine, a coordinate system transformation is performed to obtain the linear velocities $\dot{x}_0, \dot{y}_0, \dot{z}_0$ in the fixed coordinate system. The velocities are used in a Taylor series expansion to obtain the position of the center of gravity of the submarine in the x_0, y_0, z_0 system.

$$\begin{Bmatrix} x_0 \\ y_0 \\ z_0 \end{Bmatrix} (t + \Delta t) = \begin{Bmatrix} x_0 \\ y_0 \\ z_0 \end{Bmatrix} (t) + \begin{Bmatrix} \dot{x}_0 \\ \dot{y}_0 \\ \dot{z}_0 \end{Bmatrix} \cdot (\Delta t)$$

When the position of the submarine is calculated the time is advanced one increment and another iteration begins.

The iteration loop will continue to operate until some test criteria are met. For instance, if the model is performing a dive, then the model tests the value of z_0 to see if the model is at the appropriate depth. A test is also made of the pitch angle to see if it is within some specified range of zero. Lastly a check is made to ensure that all of the dive planes are at zero deflection.

II.4 CONTROL SURFACES

The motion of the model is controlled by the movement of the control surfaces. The control surfaces include the rudder, the stern planes, and the sail planes. For analysis of submarine designs the movement of the control surfaces can be governed by an automatic control system. In the model each control surface is provided with its own automatic control system. The principle of operation for all of the automatic control systems is the same. In each case the deflection of the control surface is made proportional to an error signal and the rate of change of the error. The sail planes, for instance, control the depth of the model. The calculated deflection of the planes is proportional to the depth error and the rate of change of depth.

$$\delta_c = K_1 (\text{present depth} - \text{ordered depth}) \\ + K_2 (\text{rate of change of depth})$$

A large error will initially cause a large deflection, but as the rate of change increases, the deflection will decrease until some moderate rate is achieved. Since movement of the sailplanes does not have any significant effect on any motion other than depth, its control system uses only the variables associated with depth.

The stern planes control pitch angle as well as depth. The control system accounts for this by using the pitch angle, θ , and the rate of change of θ .

$$\delta_c = K_3 (\text{present depth} - \text{ordered depth}) \\ + K_4 (\text{rate of change of depth}) \\ + K_5 (\text{pitch angle}) \\ + K_6 (\text{rate of change of pitch angle})$$

The K's in the control systems are proportionality constants known as gain. The value of the gain determines the sensitivity and response of the control system. Since the error signal continually varies during a maneuver, it is best to keep the gain low. It is desirable to have just enough gain on the error signal to get the control system moving and keep it moving in the right direction. The gain on the rate signals should be much stronger to dampen out the oscillations caused by response to the error signal.

The control surfaces move at a rate specified by the user. Whenever the calculated deflection differs from the present deflection, the control surface will move toward the calculated value at the specified rate.

CHAPTER THREE - PROGRAM USER'S GUIDE

III.1 GENERAL FEATURES

The simulation model consists of a main program and four subroutines. The four subroutines are named FUNC, RUDDER, DEPTH, and STERN. The main program reads the input data, changes units, initializes values, and prints the output. The main program performs the Taylor series expansions, the coordinate system transforms, and provides the logical statements for calling the subroutines. On the first iteration, when time equals zero, no subroutines are called; the output from this first iteration is then a statement of the initial conditions of the problem. Each succeeding iteration will call subroutine FUNC to calculate the new position of the model. Calls to the control surface subroutines are made on the basis of need, with no subroutine being called more than once in the iteration. The decision on whether or not to call RUDDER, DEPTH, or STERN is contained in a series of logical IF statements in the main program.

III.2 SUBROUTINE FUNC

Subroutine FUNC contains the six equations of motion. The

input parameters to FUNC consist of the hydrodynamic coefficients, the model velocities, the orientation in space, all of the ship's characteristics such as length and moments of inertia, the deflection of each control surface, and the propulsive coefficients. The subroutine returns the value of the six accelerations: UDT, VDT, WDT, PDT, QDT, RDT. The input and output for FUNC are all contained in COMMON /FOUR/ and COMMON /FIVE/. The solution to the matrix equation in subroutine FUNC is made possible by a call to the library function LEQTF [3]. LEQTF performs a Gaussian reduction for the matrix equation. Any similar Gaussian reduction could be substituted if LEQTF is not available. An explanation of the parameters used in the call to LEQTF is found in Appendix B.

III.3 SUBROUTINE RUDDER

Subroutine RUDDER controls all horizontal motion for the model. The input parameters include: K9, K10, T5, T6, T11, T12, T17, T18, TLAGR, RRATE, RUDAMT, COURSE, DELR, DELT, R, PSI. The subroutine returns a new value for DELR, the rudder deflection, based on a calculation using its present and ordered headings. The inputs are all contained in COMMON /THREE/ and COMMON /FIVE/. Subroutine RUDDER is called whenever the model is not on the desired course or when the rudder is deflected.

III.4 SUBROUTINE DEPTH

Subroutine DEPTH controls the forward set of dive planes. It can control either bow planes or sail planes depending on what the

submarine is fitted with. This subroutine is sensitive to the depth error and the rate of change of depth. Subroutine DEPTH will move the forward planes in the direction necessary to bring the depth error to zero. As input parameters, the subroutine uses: K1, K2, T1, T2, T13, T14, T15, T16, TLAGB, DELT, DIFF, ADIFF, ZDT, ATHETA, MAXANG, DCRIT. The subroutine returns a new value for DELB, the bow plane deflection, after each call.

If MAXANG, the maximum dive/ascent angle, has been exceeded, then DEPTH will return a diagnostic write statement. The user may specify MAXANG, the maximum pitch angle for the model. Whenever MAXANG is exceeded the diving planes are moved to reduce the pitch angle and a diagnostic signal is generated from within subroutine DEPTH. The diagnostic will say "Maximum dive/ascent angle exceeded at time _____. Standard fixup taken." Since the dive planes only begin to react when MAXANG is exceeded, the submarine will overshoot the angle before a reduction in the angle actually occurs. The diving planes will continue to operate until the pitch angle is less than MAXANG.

DEPTH makes only a minimal attempt to control the pitch angle of the model; it only warns the user when the angle is exceeded, and then moves the bow planes so that the pitch angle does not grow larger. The principal purpose of DEPTH is to provide depth control; it does this in conjunction with subroutine STERN.

III.5 SUBROUTINE STERN

Subroutine STERN controls the movement of the stern planes to

achieve the desired depth and pitch angle. The movement of the planes is sensitive to the depth error, the rate of change of depth, the pitch angle, and the rate of change of pitch. The stern planes will move to bring the depth error to zero and the pitch angle to zero. As input parameters the subroutine uses: K5, K6, K7, K8, T3, T7, T8, T10, TLAGS, STERAT, STERMX, THETA, Q, DELS, DIFF, ADIFF, ZDT, ATHETA, MAXANG, NOPICH, DCRIT. The subroutine returns a new value for the stern plane deflection, DELS. The input and output parameters are all contained in COMMON /ONE/, COMMON /FIVE/, and COMMON /SIX/. STERN is called to achieve a depth change or to correct the pitch angle.

III.6 RANGE VARIABLES

Since it is not usually possible to bring a computer simulation model to a precise depth or angle, it is necessary to define ranges around the desired depth or angle which will be acceptable to the user. For example, if the submarine were to make a depth change of 500 feet, the dive may be considered complete if the model settled within ten feet of the desired depth. The range is specified by the user to enable him to achieve whatever precision is desired. Within this range the main program will not make a call to the control surface subroutine; since no call is made, the control surfaces cannot be moved. In the program, the variable name for the depth range is NODIFF; it is usually set at ± 5 or ± 10 feet. Subroutine DEPTH will be called whenever the model is off of its ordered depth by more than NODIFF. The range about zero pitch angle is given the name NOPICH;

it is usually set at ± 1 degree. STERN is called if the pitch angle is greater than NOPICH. The variable name for achieving the proper course is ONCRS; it is usually set at ± 1 degree. RUDDER will be called whenever the model is off course by more than ONCRS.

An additional range variable is used during diving or surfacing. When the model moves within some critical range of the desired depth, it is time to begin moving the dive planes to zero angle and let the vessel glide into the desired depth. This maneuver prevents oscillation of the dive planes as the error signal and the rate signal both become small. This also ensures that the planes are at or near zero angle by the time the desired depth is reached. The name of this variable is DCRIT; it is usually set at 50, 75 or 100 feet depending on the speed of the submarine. STERN will be called whenever the model is off its ordered depth by more than DCRIT.

III.7 PROGRAM PARAMETERS

The iterative nature of the program relies on a time increment being made after each step. The size of the time increment is optional; the smaller the increment the more accurate the calculations. In selecting the size of the time increment, one must bear in mind the length of time required to complete the maneuver and the storage capacity of the program. Due to the quantity of information that is calculated by the program, it is practical to print only half of the data at one time. The other half of the data is stored in the array STOW. This data is then printed when the maneuver is complete. STOW has the capacity to hold the data resulting from 600 iterations.

A time check is incorporated in the main program to enable the user to limit the number of iterations. Achieving a number of iterations equal to the variable ICNT will cause the program to stop the present maneuver, print out all data, and see if another maneuver is desired. ICNT should not be made larger than 600 so that the available storage space is not exceeded.

Each control surface subroutine has a time lag scheme which senses the initial command to the control surface and prevents immediate action. A time lag occurs each time the direction of movement is changed. A different time lag may be specified for each control surface.

The variable INDEX is used to allow the user to perform more than one maneuver with a given submarine and then shift submarines to perform more maneuvers. If INDEX is less than or equal to zero then the same submarine coefficients will be used for each set of initial conditions. If INDEX is greater than zero the program will read new coefficients as well as a new set of initial conditions.

CHAPTER FOUR - TEST RESULTS AND CONCLUSIONS

IV.1 FUNDAMENTAL MOTION TEST

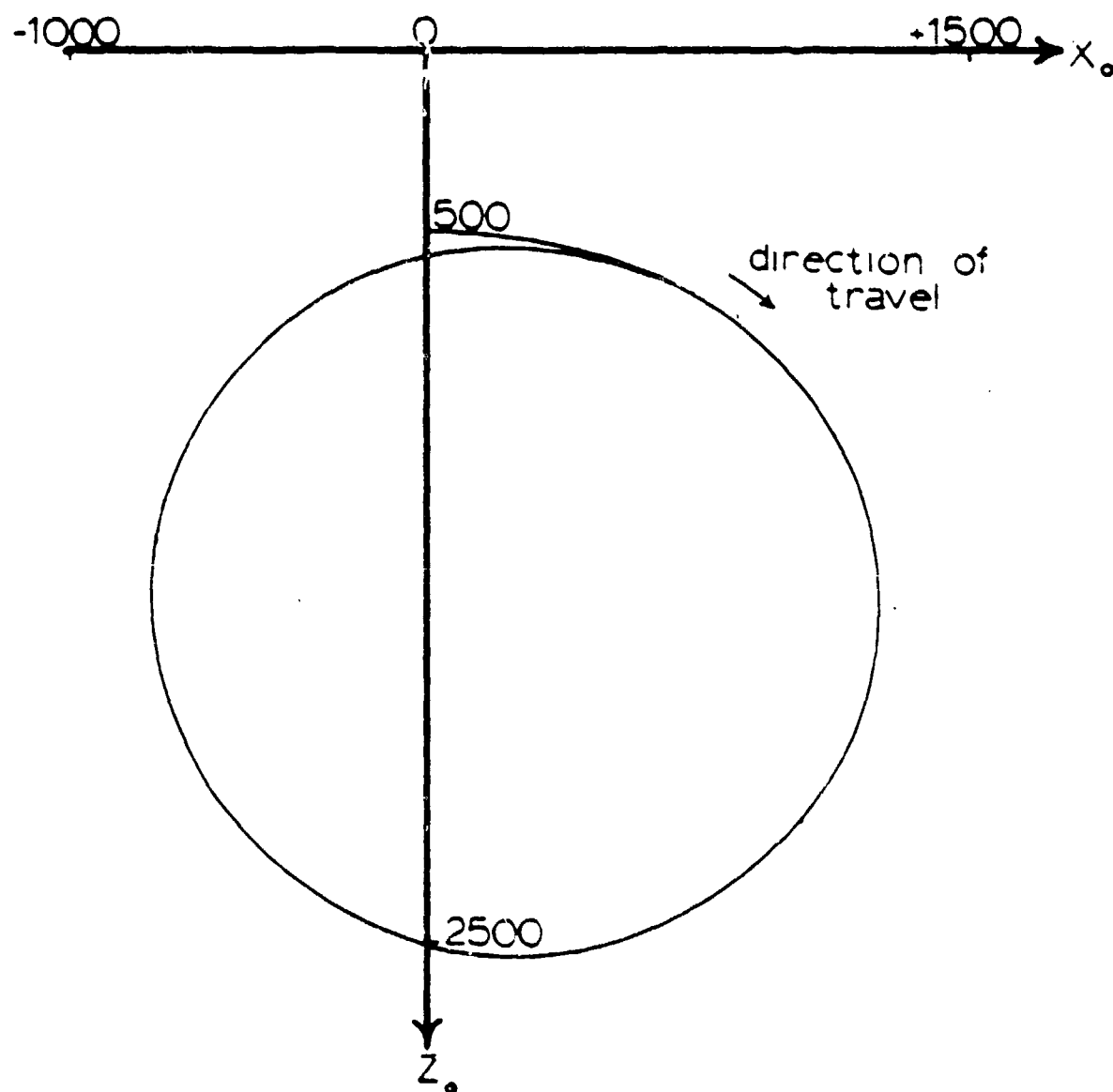
The validation of the simulation model is a matter of importance. The method used was an independent test of each component of the model, and then a series of tests with the components of the model working together. The test maneuvers were selected either because the correct dynamic response was known or because the maneuver was simple enough so that the general nature of the response could be predicted.

The first portion of the model to be tested was the subroutine FUNC. FUNC obtains the solution to the six equations of motion. It was important to establish that these equations were properly installed in the program. A test of subroutine FUNC necessitated a simultaneous test of the MAIN program to read in the hydrodynamic coefficients, set up the "A" matrix for FUNC, perform the Taylor series expansions, and write out the results. The test for FUNC was to reduce the GM of the vessel to zero and then to assume a constant angle on the dive planes. If the model was working correctly it should traverse a perfect circle in the vertical plane.

It was known that the rudder should not move and the model should not roll or yaw. The model should assume a circular trajectory with a constant angular velocity and a constant vertical component of velocity, w . Figure 3 shows the results of the maneuver. A circular trajectory was quickly achieved. The angular velocity was constant, w was constant, and the position of the model in the fixed coordinate system confirmed the circular path. The model did not roll or yaw and the rudder did not move.

IV.2 HORIZONTAL MOTION TEST

With the knowledge that the equations of motion, the Taylor series expansions, and the coordinate system transforms are operating properly, the subroutine RUDDER was the next component to test. It was decided that a simple left turn of 40 degrees would be an appropriate test. A left turn was chosen because that presents the greatest opportunity for error. The original course would be 000 degrees true and the new course would be 320 degrees true. Since the yaw angle is positive when turning right, the model must cope with a negative yaw angle as well as a proper method for dealing with course readings given in true bearing. The subroutines DEPTH and STERN were rendered inoperative to give the opportunity of seeing RUDDER operate without interference. The model would be checked to ensure that the proper rudder rate was used and that the rudder angle did not exceed the maximum angle ordered. The model would be expected to roll and squat in the turn. A change in depth was expected since the dive planes could not act. As the model approached the new heading the



Simulation Model Trajectory With Constant
Dive Plane Angle

Figure 3

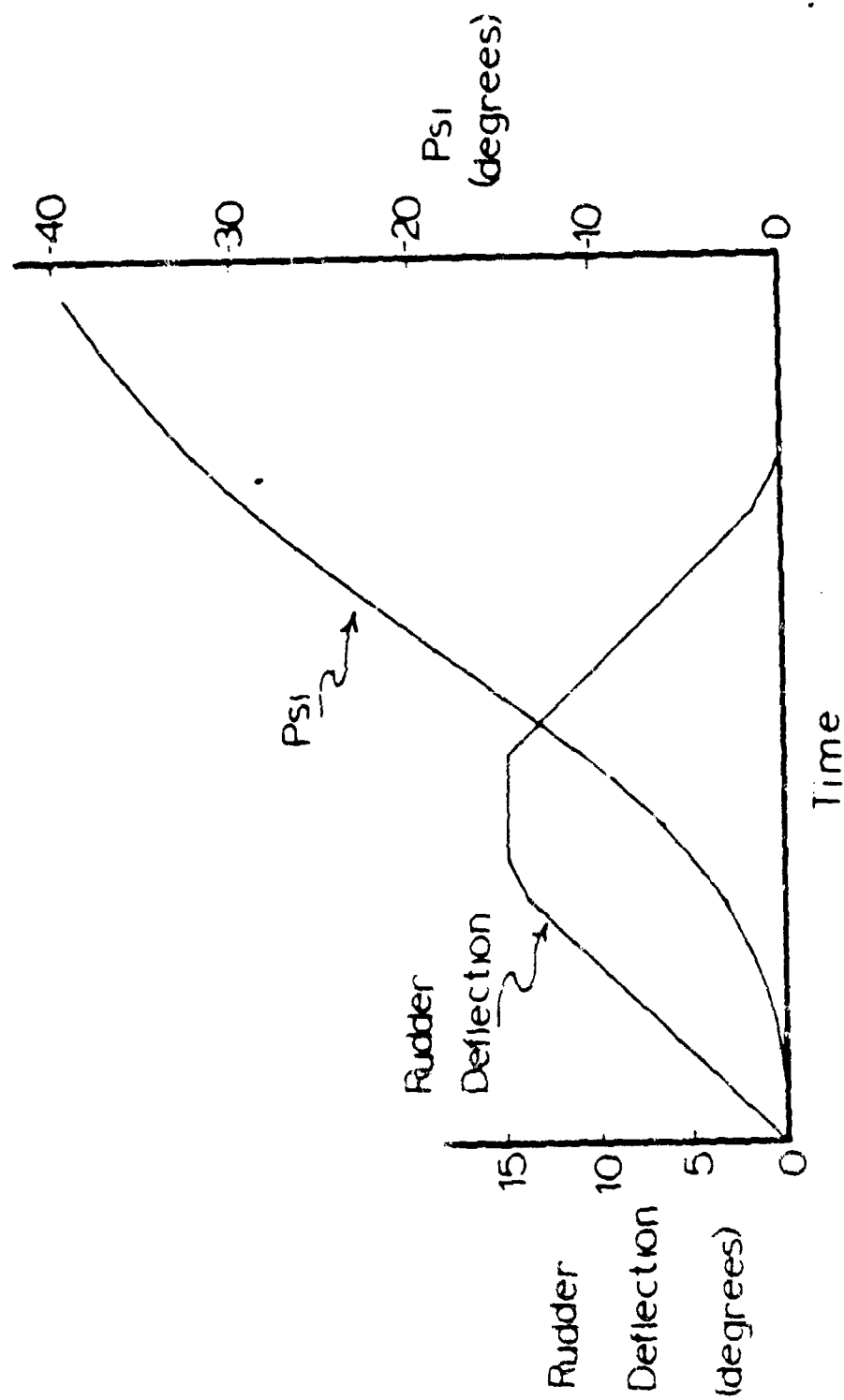
rudder should be put amidships and the vessel should steady out.

In Figure 4 the relationship between the rudder and the heading is shown as a function of time. The model reacted as predicted except possibly at the end of the turn. The rudder was amidships and the new heading was achieved but the program did not run long enough to ensure that the model was steady on the course. The model did roll, squat, and change depth as predicted. The test for RUDDER was considered to be accurate and sufficiently complete to warrant moving to the next test.

IV.3 VERTICAL MOTION TEST

Since DEPTH and STERN operate together to regulate the depth and pitch of the model, they were tested together. The test consisted of a simple dive with a depth change of 700 feet. The course was not changed so the rudder should not move. The model should not roll or yaw. The dive planes should move in the proper direction and at the ordered rate. They should not deflect more than the ordered angle. The model should pitch downward and if the maximum ordered pitch angle is exceeded then the dive planes should move to reduce the pitch angle. As it approaches the desired depth, the model should slow its rate of descent and settle within ten feet of the depth, with 0 \pm 1 degree of pitch angle.

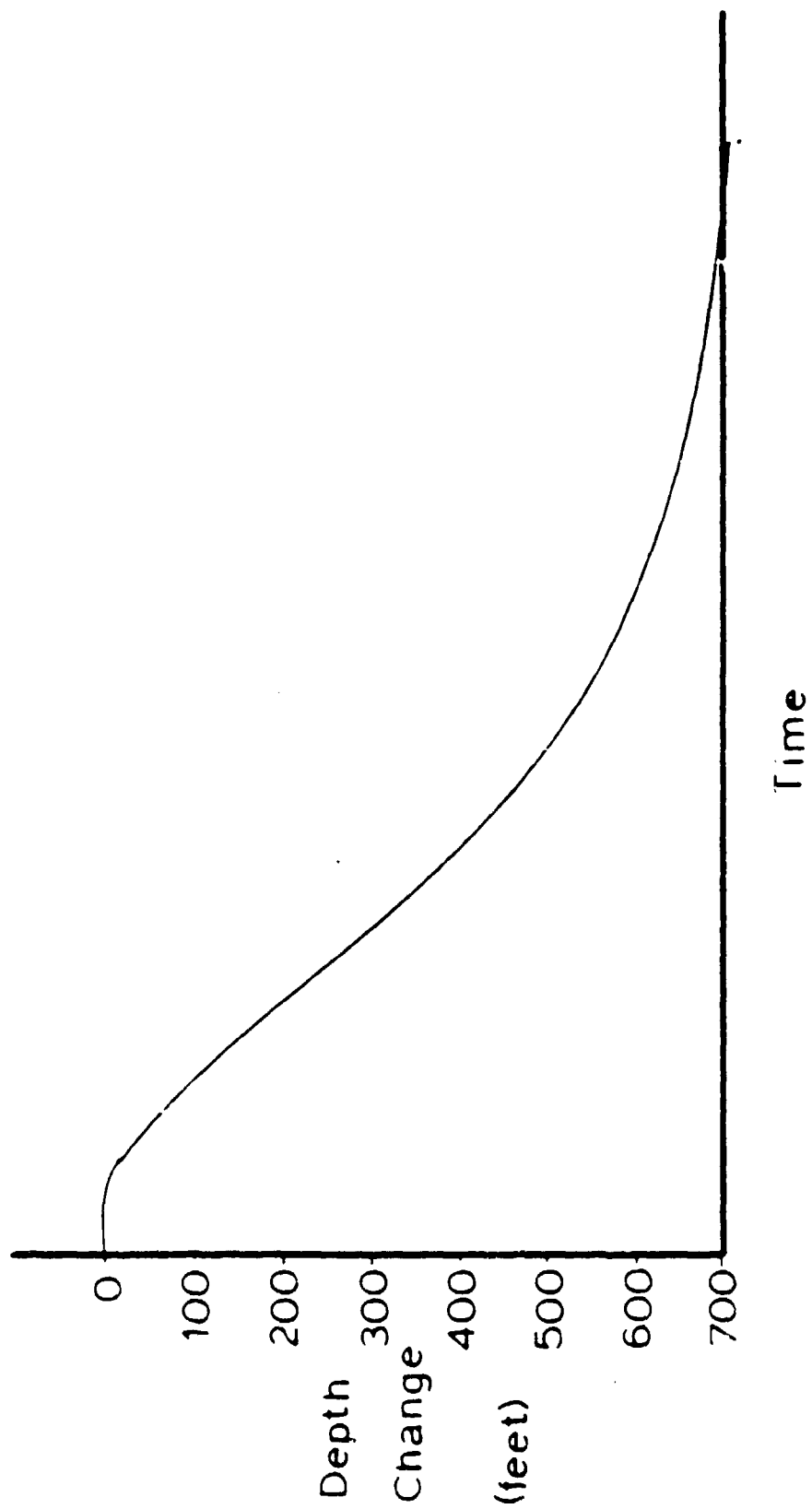
The trajectory for this test is shown in Figure 5. The model achieved steady state only six feet beyond the desired depth; the pitch angle was -.03 degrees. The dive planes were all at zero deflection. The model stayed within ten feet of the ordered depth



Rudder Deflection And Angle Of Turn As A

Function Of Time

Figure 4



Depth As A Function Of Time For A Simple Dive

Figure 5

and within one degree of zero pitch angle for twenty seconds. This was done to ensure that a steady state had been achieved.

IV.4 COMPLETE MODEL TEST

As a final test the entire simulation model must work together. This test repeated the forty degree left hand turn but this time the dive planes were allowed to react to try to maintain the depth. The program was kept running until the pitch angle was within one degree of being zero, the model was within one degree of the proper course and the depth was within ten feet of the ordered depth. The model was initialized on course 000 degrees true at an initial speed of twenty knots.

The maneuver was successfully completed. The model remained steady at 320 \pm 1 degrees true and the final depth was within one foot of the ordered depth. Both the pitch and roll angles were near one degree and were decreasing in magnitude. All of the control surfaces were at zero angle of deflection. It should be noted that the new course had been achieved during the first sixty seconds and that the course was maintained by the model while the proper depth and attitude were being obtained.

IV.5 USE AS A DESIGN TOOL

There are numerous design tasks that could be used to demonstrate the simulation model; a simple example is sufficient for this demonstration. Suppose a designer wanted to know the dynamic effects of increasing the rudder rate in a turn. The designer would elect to

use a simulation model. Since he would not want the dive planes to interfere with the analysis, he would set NODIFF, DCRIT, and NOPICH at large values so that the dive planes would not move. Then he would set RRATE, the rudder rate, at the value he desired to test and order the model to come to a new course. For this run, let's assume that the designer performed the left hand turn from 000 degrees true to 320 degrees true. He used a rudder rate of two degrees/second. He then caused the model to perform the same maneuver again with a new rudder rate of four degrees/second. With the output from these two maneuvers he could easily find the time required for the turn, the advance and transfer of the submarine, and the roll and pitch angles as a function of time. The designer could use the information for whatever analysis he had in mind. The model could be run again at new rudder rates or the dive planes could be brought into play or virtually any other maneuver could be simulated. This model can also do snap roll analysis [4].

The cost of running the simulation model is so low that it can be used on a daily basis if desired. For the test turns mentioned above, the cost was less than five dollars, which included reading the cards, compilation, execution, and 900 lines of output. A designer who used the model regularly could have the model as an on line dataset which would greatly reduce the cost of using the model.

IV.6 CONCLUSIONS AND RECOMMENDATIONS

This simulation model does accurately simulate the six degree of freedom motion of a submarine for moderate maneuvers. The cost of

operating the model is very low, which will allow frequent usage. The model is designed to permit a great deal of flexibility in the application of the model. For those designers who use the simulation model, it should be a useful design tool.

If work were continued on this simulation model, it is recommended that some time be spent in improving the appendage control subroutines. Further application of control theory would be helpful with the appendage subroutines. The data input could be organized more efficiently to remove some of the opportunity for error. It could be very useful to have a plotting routine in the model to visually display the information.

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APPENDIX A

A.1 NOTATION

Symbol	Dimensionless Form	Definition
B	$B' = \frac{B}{\rho g L^3 U^3}$	Buoyancy force, positive upward
CB		Center of buoyancy of submarine
CG		Center of mass of submarine
I_x	$I_x' = \frac{I_x}{\rho g L^5}$	Moment of inertia of submarine about x axis
I_y	$I_y' = \frac{I_y}{\rho g L^5}$	Moment of inertia of submarine about y axis
I_z	$I_z' = \frac{I_z}{\rho g L^5}$	Moment of inertia of submarine about z axis
I_{xy}	$I_{xy}' = \frac{I_{xy}}{\rho g L^5}$	Product of inertia about xy axis
I_{yz}	$I_{yz}' = \frac{I_{yz}}{\rho g L^5}$	Product of inertia about yz axes
I_{zx}	$I_{zx}' = \frac{I_{zx}}{\rho g L^5}$	Product of inertia about zx axes
K	$K' = \frac{K}{\rho g L^3 U^3}$	Hydrodynamic moment component about x axis (rolling moment)
K_0	$K_0' = \frac{K_0}{\rho g L^3 U^3}$	Rolling moment when body angle (η , δ) and control surface angles are zero
$K_{\eta\eta}$	$K_{\eta\eta}' = \frac{K_{\eta\eta}}{\rho g L^3 U^3}$	Coefficient used in representing K_0 as a function of (η -1)
K_p	$K_p' = \frac{K_p}{\rho g L^3 U^3}$	First order coefficient used in representing K as a function of p
$K_{\dot{p}}$	$K_{\dot{p}}' = \frac{K_{\dot{p}}}{\rho g L^3 U^3}$	Coefficient used in representing K as a function of \dot{p}
$K_{p p }$	$K_{p p }' = \frac{K_{p p }}{\rho g L^3 U^3}$	Second order coefficient used in representing K as a function of p
K_{pq}	$K_{pq}' = \frac{K_{pq}}{\rho g L^3 U^3}$	Coefficient used in representing K as a function of the product pq

K_{qr}	$K_{qr}' = \frac{K_{qr}}{\rho c^4}$	Coefficient used in representing K as a function of the product qr
K_r	$K_r' = \frac{K_r}{\rho c^4 U}$	First order coefficient used in representing K as a function of r
$K_{\dot{r}}$	$K_{\dot{r}}' = \frac{K_{\dot{r}}}{\rho c^5}$	Coefficient used in representing K as a function of \dot{r}
K_v	$K_v' = \frac{K_v}{\rho c^4 U}$	First order coefficient used in representing K as a function of v
$K_{\dot{v}}$	$K_{\dot{v}}' = \frac{K_{\dot{v}}}{\rho c^5}$	Coefficient used in representing K as a function of \dot{v}
$K_{v v }$	$K_{v v }' = \frac{K_{v v }}{\rho c^5}$	Second order coefficient used in representing K as a function of v
K_{vq}	$K_{vq}' = \frac{K_{vq}}{\rho c^5}$	Coefficient used in representing K as a function of the product vq
K_{vw}	$K_{vw}' = \frac{K_{vw}}{\rho c^5}$	Coefficient used in representing K as a function of the product vw
K_{wp}	$K_{wp}' = \frac{K_{wp}}{\rho c^5}$	Coefficient used in representing K as a function of the product wp
K_{wr}	$K_{wr}' = \frac{K_{wr}}{\rho c^5}$	Coefficient used in representing K as a function of the product wr
$K_{\delta r}$	$K_{\delta r}' = \frac{K_{\delta r}}{\rho c^4 U}$	First order coefficient used in representing K as a function of δ_r
L	$L' = L$	Overall length of submarine
m	$m' = \frac{m}{\rho c^3}$	Mass of submarine, including water in free-flooding spaces
M	$M' = \frac{M}{\rho c^3 U^2}$	Hydrodynamic moment component about y axis (pitching moment)
M_0	$M_0' = \frac{M_0}{\rho c^3 U^2}$	Pitching moment when body angles (γ, β) and control surface angles are zero
M_{pp}	$M_{pp}' = \frac{M_{pp}}{\rho c^4}$	Second order coefficient used in representing M as a function of p . First order coefficient is zero.
M_q	$M_q' = \frac{M_q}{\rho c^4 U}$	First order coefficient used in representing M as a function of q
$M_{\dot{q}}$	$M_{\dot{q}}' = \frac{M_{\dot{q}}}{\rho c^5 U}$	First order coefficient used in representing M_q as a function of $(\dot{q}-1)$
$M_{\dot{q}}$	$M_{\dot{q}}' = \frac{M_{\dot{q}}}{\rho c^5}$	Coefficient used in representing M as a function of \dot{q}

$M_{q q}$	$M_{q q}' = \frac{M_{q q}}{2\rho c^2}$	Second order coefficient used in representing M as a function of q
$M_{ q \delta a}$	$M_{ q \delta a}' = \frac{M_{ q \delta a}}{2\rho c^2 U}$	Coefficient used in representing $M_{\delta a}$ as a function q
M_{rp}	$M_{rp}' = \frac{M_{rp}}{2\rho c^2}$	Coefficient used in representing M as a function of the product rp
M_{rr}	$M_{rr}' = \frac{M_{rr}}{2\rho c^2}$	Second order coefficient used in representing M as a function of r . First order coefficient is zero
M_{vp}	$M_{vp}' = \frac{M_{vp}}{2\rho c^2}$	Coefficient used in representing M as a function of the product vp
M_{vr}	$M_{vr}' = \frac{M_{vr}}{2\rho c^2}$	Coefficient used in representing M as a function of the product vr
M_{vv}	$M_{vv}' = \frac{M_{vv}}{2\rho c^2}$	Second order coefficient used in representing M as a function of v
M_w	$M_w' = \frac{M_w}{2\rho c^2 U}$	First order coefficient used in representing M as a function of w
$M_{w\eta}$	$M_{w\eta}' = \frac{M_{w\eta}}{2\rho c^2 U}$	First order coefficient used in representing M_w as a function of $(\eta-1)$
M_{ω}	$M_{\omega}' = \frac{M_{\omega}}{2\rho c^2}$	Coefficient used in representing M as a function of ω
$M_{ \omega }$	$M_{ \omega }' = \frac{M_{ \omega }}{2\rho c^2 U}$	First order coefficient used in representing M as a function of $ \omega $; equal to zero for symmetrical function
$M_{ \omega q}$	$M_{ \omega q}' = \frac{M_{ \omega q}}{2\rho c^2}$	Coefficient used in representing M_q as a function of ω
$M_{\omega \omega }$	$M_{\omega \omega }' = \frac{M_{\omega \omega }}{2\rho c^2}$	Second order coefficient used in representing M as a function of ω
$M_{\omega \omega \eta}$	$M_{\omega \omega \eta}' = \frac{M_{\omega \omega \eta}}{2\rho c^2}$	First order coefficient used in representing $M_{\omega \omega }$ as a function of $(\eta-1)$
$M_{\omega\omega}$	$M_{\omega\omega}' = \frac{M_{\omega\omega}}{2\rho c^2}$	Second order coefficient used in representing M as a function of ω ; equal to zero for symmetrical function
$M_{\delta b}$	$M_{\delta b}' = \frac{M_{\delta b}}{2\rho c^2 U^2}$	First order coefficient used in representing M as a function of δ_b
$M_{\delta a}$	$M_{\delta a}' = \frac{M_{\delta a}}{2\rho c^2 U^2}$	First order coefficient used in representing M as a function of δ_a
$M_{\delta a\eta}$	$M_{\delta a\eta}' = \frac{M_{\delta a\eta}}{2\rho c^2 U^2}$	First order coefficient used in representing $M_{\delta a}$ as a function of $(\eta-1)$

N	$N = \frac{N}{\rho c^2 U}$	Hydrodynamic moment component about z axis (yawing moment)
N_0	$N_0 = \frac{N_0}{\rho c^2 U}$	Yawing moment when body angles (α, β) and control surface angles are zero
N_p	$N_p = \frac{N_p}{\rho c^2 U}$	First order coefficient used in representing N as a function of p
$N_{\dot{p}}$	$N_{\dot{p}} = \frac{N_{\dot{p}}}{\rho c^2}$	Coefficient used in representing N as a function of \dot{p}
N_{pq}	$N_{pq} = \frac{N_{pq}}{\rho c^2}$	Coefficient used in representing N as a function of the product pq
N_{qr}	$N_{qr} = \frac{N_{qr}}{\rho c^2}$	Coefficient used in representing N as a function of the product qr
N_r	$N_r = \frac{N_r}{\rho c^2 U}$	First order coefficient used in representing N as a function of r
$N_{r\eta}$	$N_{r\eta} = \frac{N_{r\eta}}{\rho c^2 U}$	First order coefficient used in representing N_r as a function of $(\eta-1)$
$N_{\dot{r}}$	$N_{\dot{r}} = \frac{N_{\dot{r}}}{\rho c^2}$	Coefficient used in representing N as a function of \dot{r}
$N_{r r }$	$N_{r r } = \frac{N_{r r }}{\rho c^2}$	Second order coefficient used in representing N as a function of r
$N_{r \dot{r} }$	$N_{r \dot{r} } = \frac{N_{r \dot{r} }}{\rho c^2 U}$	Coefficient used in representing $N_{r r }$ as a function of r
N_v	$N_v = \frac{N_v}{\rho c^2 U}$	First order coefficient used in representing N as a function of v
$N_{v\eta}$	$N_{v\eta} = \frac{N_{v\eta}}{\rho c^2 U}$	First order coefficient used in representing N_v as a function of $(\eta-1)$
$N_{\dot{v}}$	$N_{\dot{v}} = \frac{N_{\dot{v}}}{\rho c^2}$	Coefficient used in representing N as a function of \dot{v}
N_{vq}	$N_{vq} = \frac{N_{vq}}{\rho c^2}$	Coefficient used in representing N as a function of the product vq
$N_{v r }$	$N_{v r } = \frac{N_{v r }}{\rho c^2}$	Coefficient used in representing N_r as a function of v
$N_{v \dot{v} }$	$N_{v \dot{v} } = \frac{N_{v \dot{v} }}{\rho c^2}$	Second order coefficient used in representing N as a function of v
$N_{v \dot{v} \eta}$	$N_{v \dot{v} \eta} = \frac{N_{v \dot{v} \eta}}{\rho c^2}$	First order coefficient used in representing $N_{v \dot{v} }$ as a function of $(\eta-1)$

N_{vw}	$N_{vw}' = \frac{N_{vw}}{\rho c^3}$	Coefficient used in representing N as a function of the product vw
N_{wp}	$N_{wp}' = \frac{N_{wp}}{\rho c^3}$	Coefficient used in representing N as a function of the product wp
N_{wr}	$N_{wr}' = \frac{N_{wr}}{\rho c^3}$	Coefficient used in representing N as a function of the product wr
$N_{\delta r}$	$N_{\delta r}' = \frac{N_{\delta r}}{\rho c^3 U}$	First order coefficient used in representing N as a function of δr
$N_{\delta \eta}$	$N_{\delta \eta}' = \frac{N_{\delta \eta}}{\rho c^3 U}$	First order coefficient used in representing $N_{\delta r}$ as a function of $(\eta - 1)$
p	$p' = \frac{p c}{U}$	Angular velocity component about x axis relative to fluid (roll)
\dot{p}	$\dot{p}' = \frac{\dot{p} c^2}{U^2}$	Angular acceleration component about x axis relative to fluid
q	$q' = \frac{q c}{U}$	Angular velocity component about y axis relative to fluid (pitch)
\dot{q}	$\dot{q}' = \frac{\dot{q} c^2}{U^2}$	Angular acceleration component about y axis relative to fluid
r	$r' = \frac{r c}{U}$	Angular velocity component about z axis relative to fluid (yaw)
\dot{r}	$\dot{r}' = \frac{\dot{r} c^2}{U^2}$	Angular acceleration component about z axis relative to fluid
U	$U' = \frac{U}{U}$	Linear velocity of origin of body axes relative to fluid
u	$u' = \frac{u}{U}$	Component of U in direction of the x axis
\dot{u}	$\dot{u}' = \frac{\dot{u} c}{U^2}$	Time rate of change of u in direction of the x axis
u_c	$u_c' = \frac{u_c}{U}$	Command speed: steady value of ahead speed component u for a given propeller rpm when body angles α , β and control surface angles are zero. Sign changes with propeller reversal
v	$v' = \frac{v}{U}$	Component of U in direction of the y axis
\dot{v}	$\dot{v}' = \frac{\dot{v} c}{U^2}$	Time rate of change of v in direction of the y axis

w	$w' = \frac{w}{U}$	Component of U in direction of the z axis
\dot{w}	$\dot{w}' = \frac{\dot{w}}{U^2}$	Time rate of change of w in direction of the z axis
W	$W' = \frac{W}{\rho \ell^3 U^2}$	Weight, including water in free flooding spaces
x	$x' = \frac{x}{\ell}$	Longitudinal body axis; also the coordinate of a point relative to the origin of body axes
x_B	$x_B' = \frac{x_B}{\ell}$	The x coordinate of CB
x_C	$x_C' = \frac{x_C}{\ell}$	The x coordinate of CG
x_0	$x_0' = \frac{x_0}{\ell}$	A coordinate of the displacement of CG relative to the origin of a set of fixed axes
X	$X' = \frac{X}{\rho \ell^2 U^2}$	Hydrodynamic force component along x axis (longitudinal, or axial, force)
X_{qq}	$X_{qq}' = \frac{X_{qq}}{\rho \ell^2}$	Second order coefficient used in representing X as a function of q . First order coefficient is zero
X_{rp}	$X_{rp}' = \frac{X_{rp}}{\rho \ell^2}$	Coefficient used in representing X as a function of the product rp
X_{rr}	$X_{rr}' = \frac{X_{rr}}{\rho \ell^2}$	Second order coefficient used in representing X as a function of r . First order coefficient is zero
$X_{\dot{u}}$	$X_{\dot{u}}' = \frac{X_{\dot{u}}}{\rho \ell^2}$	Coefficient used in representing X as a function of \dot{u}
X_{uu}	$X_{uu}' = \frac{X_{uu}}{\rho \ell^2}$	Second order coefficient used in representing X as a function of u in the non-propelled case. First order coefficient is zero
X_{vr}	$X_{vr}' = \frac{X_{vr}}{\rho \ell^2}$	Coefficient used in representing X as a function of the product vr
X_{vv}	$X_{vv}' = \frac{X_{vv}}{\rho \ell^2}$	Second order coefficient used in representing X as a function of v . First order coefficient is zero
$X_{v\eta}$	$X_{v\eta}' = \frac{X_{v\eta}}{\rho \ell^2}$	First order coefficient used in representing X_{vv} as a function of $(\eta - 1)$
X_{wq}	$X_{wq}' = \frac{X_{wq}}{\rho \ell^2}$	Coefficient used in representing X as a function of the product wq

X_{ww}	$X_{ww}' = \frac{X_{ww}}{\rho L^2}$	Second order coefficient used in representing X as a function of w . First order coefficient is zero
$X_{w\eta}$	$X_{w\eta}' = \frac{X_{w\eta}}{\rho L^2}$	First order coefficient used in representing X_{ww} as a function of $(\eta-1)$
$X_{\delta b \delta b}$	$X_{\delta b \delta b}' = \frac{X_{\delta b \delta b}}{\rho L^2 U^2}$	Second order coefficient used in representing X as a function of δ_b . First order coefficient is zero
$X_{\delta r \delta r}$	$X_{\delta r \delta r}' = \frac{X_{\delta r \delta r}}{\rho L^2 U^2}$	Second order coefficient used in representing X as a function of δ_r . First order coefficient is zero
$X_{\delta r \delta \eta}$	$X_{\delta r \delta \eta}' = \frac{X_{\delta r \delta \eta}}{\rho L^2 U^2}$	First order coefficient used in representing $X_{\delta r \delta r}$ as a function of $(\eta-1)$
$X_{\delta s \delta s}$	$X_{\delta s \delta s}' = \frac{X_{\delta s \delta s}}{\rho L^2 U^2}$	Second order coefficient used in representing X as a function of δ_s . First order coefficient is zero
$X_{\delta s \delta \eta}$	$X_{\delta s \delta \eta}' = \frac{X_{\delta s \delta \eta}}{\rho L^2 U^2}$	First order coefficient used in representing $X_{\delta s \delta s}$ as a function of $(\eta-1)$
y	$y' = \frac{y}{L}$	Lateral body axis; also the coordinate of a point relative to the origin of body axes
y_B	$y_B' = \frac{y_B}{L}$	The y coordinate of CB
y_G	$y_G' = \frac{y_G}{L}$	The y coordinate of CG
y_0	$y_0' = \frac{y_0}{L}$	A coordinate of the displacement of CG relative to the origin of a set of fixed axes
Y	$Y' = \frac{Y}{\rho L^2 U^2}$	Hydrodynamic force component along y axis (lateral force)
Y_0	$Y_0' = \frac{Y}{\rho L^2 U^2}$	Lateral force when body angles (α, β) and control surface angles are zero
Y_p	$Y_p' = \frac{Y_p}{\rho L^2 U}$	First order coefficient used in representing Y as a function of p
$Y_{\dot{p}}$	$Y_{\dot{p}}' = \frac{Y_{\dot{p}}}{\rho L^2}$	Coefficient used in representing Y as a function of \dot{p}
$Y_{p p }$	$Y_{p p }' = \frac{Y_{p p }}{\rho L^2}$	Second order coefficient used in representing Y as a function of p

Y_{pq}	$Y_{pq}' = \frac{Y_{pq}}{20C^2}$	Coefficient used in representing Y as a function of the product pq
Y_{qr}	$Y_{qr}' = \frac{Y_{qr}}{20C^2}$	Coefficient used in representing Y as a function of the product qr
Y_r	$Y_r' = \frac{Y_r}{20C^2U}$	First order coefficient used in representing Y as a function of r
$Y_{r\eta}$	$Y_{r\eta}' = \frac{Y_{r\eta}}{20C^2U}$	First order coefficient used in representing Y_r as a function of $(\eta-1)$
Y_t	$Y_t' = \frac{Y_t}{20C^2}$	Coefficient used in representing Y as a function of t
$Y_{r sr}$	$Y_{r sr}' = \frac{Y_{r sr}}{20C^2U}$	Coefficient used in representing Y_{sr} as a function of r
Y_v	$Y_v' = \frac{Y_v}{20C^2U}$	First order coefficient used in representing Y as a function of v
$Y_{v\eta}$	$Y_{v\eta}' = \frac{Y_{v\eta}}{20C^2U}$	First order coefficient used in representing Y_v as a function of $(\eta-1)$
$Y_{\dot{v}}$	$Y_{\dot{v}}' = \frac{Y_{\dot{v}}}{20C^2}$	Coefficient used in representing Y as a function of \dot{v}
Y_{vq}	$Y_{vq}' = \frac{Y_{vq}}{20C^2}$	Coefficient used in representing Y as a function of the product vq
$Y_{v r }$	$Y_{v r }' = \frac{Y_{v r }}{20C^2}$	Coefficient used in representing Y_v as a function of r
$Y_{v v }$	$Y_{v v }' = \frac{Y_{v v }}{20C^2}$	Second order coefficient used in representing Y as a function of v
$Y_{v v \eta}$	$Y_{v v \eta}' = \frac{Y_{v v \eta}}{20C^2}$	First order coefficient used in representing $Y_{v v }$ as a function of $(\eta-1)$
Y_{vw}	$Y_{vw}' = \frac{Y_{vw}}{20C^2}$	Coefficient used in representing Y as a function of the product vw
Y_{wp}	$Y_{wp}' = \frac{Y_{wp}}{20C^2}$	Coefficient used in representing Y as a function of the product wp
Y_{wr}	$Y_{wr}' = \frac{Y_{wr}}{20C^2}$	Coefficient used in representing Y as a function of the product wr
Y_{sr}	$Y_{sr}' = \frac{Y_{sr}}{20C^2U^2}$	First order coefficient used in representing Y as a function of sr
$Y_{sr\eta}$	$Y_{sr\eta}' = \frac{Y_{sr\eta}}{20C^2U^2}$	First order coefficient used in representing Y_{sr} as a function of $(\eta-1)$

z	$z' = \frac{z}{l}$	Normal body axis; also the coordinate of a point relative to the origin of body axes
z_B	$z_B' = \frac{z_B}{l}$	The z coordinate of CB
z_G	$z_G' = \frac{z_G}{l}$	The z coordinate of CG
z_o	$z_o' = \frac{z_o}{l}$	A coordinate of the displacement of CG relative to the origin of a set of fixed axes
Z	$Z' = \frac{Z}{\rho c^2 U^2}$	Hydrodynamic force component along z axis (normal force)
Z_o	$Z_o' = \frac{Z_o}{\rho c^2 U^2}$	Normal force when body angles (α, β) and control surface angles are zero
Z_{pp}	$Z_{pp}' = \frac{Z_{pp}}{\rho c^2}$	Second order coefficient used in representing Z as a function of p . First order coefficient is zero
Z_q	$Z_q' = \frac{Z_q}{\rho c^2 U}$	First order coefficient used in representing Z as a function of q
$Z_{q\eta}$	$Z_{q\eta}' = \frac{Z_{q\eta}}{\rho c^2 U}$	First order coefficient used in representing Z_q as a function of $(\eta-1)$
$Z_{\dot{q}}$	$Z_{\dot{q}}' = \frac{Z_{\dot{q}}}{\rho c^2}$	Coefficient used in representing Z as a function of \dot{q}
$Z_{ q \beta}$	$Z_{ q \beta}' = \frac{Z_{ q \beta}}{\rho c^2 U}$	Coefficient used in representing $Z_{ q }$ as a function of β
Z_{rp}	$Z_{rp}' = \frac{Z_{rp}}{\rho c^2}$	Coefficient used in representing Z as a function of the product rp
Z_{rr}	$Z_{rr}' = \frac{Z_{rr}}{\rho c^2}$	Second order coefficient used in representing Z as a function of r . First order coefficient is zero
Z_w	$Z_w' = \frac{Z_w}{\rho c^2 U}$	First order coefficient used in representing Z as a function of w
$Z_{w\eta}$	$Z_{w\eta}' = \frac{Z_{w\eta}}{\rho c^2 U}$	First order coefficient used in representing Z_w as a function of $(\eta-1)$
$Z_{\dot{w}}$	$Z_{\dot{w}}' = \frac{Z_{\dot{w}}}{\rho c^2}$	Coefficient used in representing Z as a function of \dot{w}
$Z_{ w }$	$Z_{ w }' = \frac{Z_{ w }}{\rho c^2 U}$	First order coefficient used in representing Z as a function of $ w $; equal to zero for symmetrical function
$Z_{w q }$	$Z_{w q }' = \frac{Z_{w q }}{\rho c^2}$	Coefficient used in representing Z_w as a function of q

$Z_{w w }$	$Z_{w w }' = \frac{Z_{w w }}{2\rho c^2}$	Second order coefficient used in representing Z as a function of w
$Z_{w w \eta}$	$Z_{w w \eta}' = \frac{Z_{w w \eta}}{2\rho c^2}$	First order coefficient used in representing $Z_{w w }$ as a function of $(\eta-1)$
Z_{ww}	$Z_{ww}' = \frac{Z_{ww}}{2\rho c^2}$	Second order coefficient used in representing Z as a function of w ; equal to zero for symmetrical function
$Z_{\delta b}$	$Z_{\delta b}' = \frac{Z_{\delta b}}{2\rho c^2 U^2}$	First order coefficient used in representing Z as a function of δ_b
$Z_{\delta s}$	$Z_{\delta s}' = \frac{Z_{\delta s}}{2\rho c^2 U^2}$	First order coefficient used in representing Z as a function of δ_s
$Z_{\delta \eta}$	$Z_{\delta \eta}' = \frac{Z_{\delta \eta}}{2\rho c^2 U^2}$	First order coefficient used in representing $Z_{\delta s}$ as a function of $(\eta-1)$
α		Angle of attack
β		Angle of drift
δ_b		Deflection of bowplane or sailplane
δ_r		Deflection of rudder
δ_s		Deflection of sternplane
η		The ratio $\frac{u_c}{U}$
θ		Angle of pitch
ϕ		Angle of yaw
ψ		Angle of roll
a_1, b_1, c_1		Sets of constants used in the representation of propeller thrust in the axial equation

A.2 EQUATIONS OF MOTION
AXIAL FORCE

$$\begin{aligned}
 m \left[\dot{u} - vr + wq - x_G (q^2 + r^2) + y_G (pq - \dot{r}) + z_G (pr + \dot{q}) \right] = \\
 + \frac{\rho}{2} l^4 \left[X_{qq}' q^2 + X_{rr}' r^2 + X_{rp}' rp \right] \\
 + \frac{\rho}{2} l^3 \left[X_{\dot{u}}' \dot{u} + X_{vr}' vr + X_{wq}' wq \right] \\
 + \frac{\rho}{2} l^3 \left[X_{uu}' u^2 + X_{vv}' v^2 + X_{ww}' w^2 \right] \\
 + \frac{\rho}{2} l^2 u^2 \left[X_{\delta r \delta r}' \delta r^2 + X_{\delta s \delta s}' \delta s^2 + X_{\delta b \delta b}' \delta b^2 \right] \\
 + \frac{1}{2} \rho l^2 \left[a_i u^2 + b_i uu_c + c_i u_c^2 \right] \\
 - (W - B) \sin \theta \\
 + \frac{\rho}{2} l^2 \left[X_{vv\eta}' v^2 + X_{ww\eta}' w^2 + X_{\delta r \delta r \eta}' \delta r^2 u^2 \right. \\
 \left. + X_{\delta s \delta s \eta}' \delta s^2 u^2 \right] (\eta-1)
 \end{aligned}$$

LATERAL FORCE

$$\begin{aligned}
& m \left[\dot{v} - wp + ur - y_G (r^2 + p^2) + z_G (qr - \dot{p}) + x_G (qp + \dot{r}) \right] = \\
& + \frac{\rho}{2} L^3 \left[Y_{\dot{r}}' \dot{r} + Y_{\dot{p}}' \dot{p} + Y_{p|p|}' p|p| + Y_{pq}' pq + Y_{qr}' qr \right] \\
& + \frac{\rho}{2} L^3 \left[Y_{\dot{v}}' \dot{v} + Y_{vq}' vq + Y_{wp}' wp + Y_{wr}' wr \right] \\
& + \frac{\rho}{2} L^3 \left[Y_{\dot{r}}' ur + Y_{\dot{p}}' up + Y_{|r|\delta r}' u|r|\delta r + Y_{v|r|}' \frac{v}{|v|} |(v^2 + w^2)^{\frac{1}{2}}| |r| \right] \\
& + \frac{\rho}{2} L^3 \left[Y_{\dot{u}}' u^2 + Y_{uv}' uv + Y_{v|v|}' v |(v^2 + w^2)^{\frac{1}{2}}| \right] \\
& + \frac{\rho}{2} L^3 \left[Y_{vw}' vw + Y_{\delta r}' u^2 \delta r \right] \\
& + (W - B) \cos \theta \sin \phi \\
& + \frac{\rho}{2} L^3 Y_{r\eta}' ur (\eta - 1) \\
& + \frac{\rho}{2} L^3 \left[Y_{v\eta}' uv + Y_{v|v|\eta}' v |(v^2 + w^2)^{\frac{1}{2}}| + Y_{\delta r\eta}' \delta r u^2 \right] (\eta - 1)
\end{aligned}$$

NORMAL FORCE

$$\begin{aligned}
& m \left[\dot{w} - uq + vp - z_G (p^2 + q^2) + x_G (rp - \dot{q}) + y_G (rq + \dot{p}) \right] = \\
& + \frac{\rho}{2} \ell^4 \left[Z_{\dot{q}}' \dot{q} + Z_{pp}' p^2 + Z_{rr}' r^2 + Z_{rp}' rp \right] \\
& + \frac{\rho}{2} \ell^3 \left[Z_{\dot{w}}' \dot{w} + Z_{vr}' vr + Z_{vp}' vp \right] \\
& + \frac{\rho}{2} \ell^3 \left[Z_q' uq + Z_{|q|\delta s}' u|q|\delta s + Z_{w|q|}' \frac{w}{|w|} |(v^2 + w^2)^{\frac{1}{2}}| |q| \right] \\
& + \frac{\rho}{2} \ell^3 \left[Z_u' u^2 + Z_w' uw + Z_{w|w|}' w |(v^2 + w^2)^{\frac{1}{2}}| \right] \\
& + \frac{\rho}{2} \ell^3 \left[Z_{|w|}' u|w| + Z_{ww}' |w| (v^2 + w^2)^{\frac{1}{2}} \right] \\
& + \frac{\rho}{2} \ell^3 \left[Z_{vv}' v^2 + Z_{\delta s}' u^2 \delta s + Z_{\delta b}' u^2 \delta b \right] \\
& + (W - B) \cos \theta \cos \phi \\
& + \frac{\rho}{2} \ell^3 Z_{q\eta}' uq (\eta - 1) \\
& + \frac{\rho}{2} \ell^3 \left[Z_{uw}' uw + Z_{w|w|\eta}' w |(v^2 + w^2)^{\frac{1}{2}}| + Z_{\delta s\eta}' \delta s u^2 \right] (\eta - 1)
\end{aligned}$$

ROLLING MOMENT

$$\begin{aligned}
& I_x \dot{p} + (I_z - I_y) qr - (\dot{r} + pq) I_{xz} + (r^2 - q^2) I_{yz} + (pr - \dot{q}) I_{xy} \\
& + m \left[y_G (\dot{w} - uq + vp) - z_G (\dot{v} - wp + ur) \right] \\
& + \frac{\rho}{2} I^0 \left[K_p' \dot{p} + K_r' \dot{r} + K_{qr}' qr + K_{pq}' pq + K_{p|p|}' p|p| \right] \\
& + \frac{\rho}{2} I^4 \left[K_p' up + K_r' ur + K_v' \dot{v} \right] \\
& + \frac{\rho}{2} I^4 \left[K_{vq}' vq + K_{wp}' wp + K_{wr}' wr \right] \\
& + \frac{\rho}{2} I^3 \left[K_u' u^2 + K_v' uv + K_v|v|' v|(v^2 + w^2)^{\frac{1}{2}} \right] \\
& + \frac{\rho}{2} I^3 \left[K_{vw}' vw + K_{\delta r}' u^2 \delta r \right] \\
& + (y_G W - y_B B) \cos \theta \cos \phi - (z_G W - z_B B) \cos \theta \sin \phi \\
& + \frac{\rho}{2} I^3 K_{\eta}' u^2 (\eta - 1)
\end{aligned}$$

PITCHING MOMENT

$$I_y \dot{q} + (I_x - I_z) rp - (\dot{p} + qr) I_{xy} + (p^2 - r^2) I_{zx} + (qp - \dot{r}) I_{yz}$$

$$+ m \left[z_G (\dot{u} - vr + wq) - x_G (\dot{w} - uq + vp) \right] =$$

$$+ \frac{\partial}{\partial t} I^0 \left[M_q' \dot{q} + M_{pp}' p^2 + M_{rr}' r^2 + M_{rp}' rp + M_{q|q|}' |q| |q| \right]$$

$$+ \frac{\partial}{\partial t} I^0 \left[M_{\dot{w}}' \dot{w} + M_{vr}' vr + M_{vp}' vp \right]$$

$$+ \frac{\partial}{\partial t} I^0 \left[M_q' u + M_{|q|\delta s}' u |q| \delta s + M_{|w|q|}' |w| |q| (v^2 + w^2)^{\frac{1}{2}} |q| \right]$$

$$+ \frac{\partial}{\partial t} I^0 \left[M_u' u^2 + M_{uw}' uw + M_{w|w|}' |w| (v^2 + w^2)^{\frac{1}{2}} |w| \right]$$

$$+ \frac{\partial}{\partial t} I^0 \left[M_{|w|}' u |w| + M_{ww}' |w| (v^2 + w^2)^{\frac{1}{2}} |w| \right]$$

$$+ \frac{\partial}{\partial t} I^0 \left[M_{vv}' v^2 + M_{\delta s}' u^2 \delta s + M_{\delta b}' u^2 \delta b \right]$$

$$- (x_G W - x_B B) \cos \theta \cos \phi - (z_G W - z_B B) \sin \theta$$

$$+ \frac{\partial}{\partial t} I^0 M_{q\eta}' u q (\eta - 1)$$

$$+ \frac{\partial}{\partial t} I^0 \left[M_{w\eta}' uw + M_{w|w|\eta}' |w| (v^2 + w^2)^{\frac{1}{2}} |w| + M_{\delta s\eta}' \delta s u^2 (\eta - 1) \right]$$

YAWING MOMENT

$$I_z \dot{r} + (I_y - I_x) pq - (\dot{q} + rp) I_{yz} + (q^2 - p^2) I_{xy} + (rq - \dot{p}) I_{zx}$$

$$+ m [x_G (\dot{v} - wp + ur) - y_G (\dot{u} - vr + wq)] =$$

$$+ \frac{\rho}{2} L^6 [N_r' \dot{r} + N_p' \dot{p} + N_{pq}' pq + N_{qr}' qr + N_{r|r}' |r|r]$$

$$+ \frac{\rho}{2} L^6 [N_v' \dot{v} + N_{wr}' wr + N_{wp}' wp + N_{vq}' vq]$$

$$+ \frac{\rho}{2} L^4 [N_p' up + N_r' ur + N_{|r|\delta r}' u|r|\delta r + N_{|v|r}' |(v^2 + w^2)^{\frac{1}{2}}|r]$$

$$+ \frac{\rho}{2} L^3 [N_u' u^2 + N_v' uv + N_{v|v}' v|(v^2 + w^2)^{\frac{1}{2}}]$$

$$+ \frac{\rho}{2} L^3 [N_{vw}' vw + N_{\delta r}' u^2 \delta r]$$

$$+ (x_G W - x_B B) \cos \theta \sin \theta + (y_G W - y_B B) \sin \theta$$

$$+ \frac{\rho}{2} L^4 N_{r\eta}' ur (\eta - 1)$$

$$+ \frac{\rho}{2} L^3 [N_{v\eta}' uv + N_{v|v|\eta}' v|(v^2 + w^2)^{\frac{1}{2}}| + N_{\delta r\eta}' \delta r u^2] (\eta - 1)$$

A.3 AXIS TRANSFORMATIONS [2]

1) A transformation of an axes system takes a quantity described in one frame of reference and transforms it into another frame of reference such that if we measured the same quantity in the second frame of reference the transformed quantity and the measured quantity would be identical.

2) Transforms between frames are needed in the study of the motions of ocean vehicles because the equations of motion for such a vehicle are most easily derived in the inertial frame attached to the earth (x_0, y_0, z_0) frame, while the forces acting on the vehicle are most easily evaluated in the frame attached to the vehicle (x, y, z). Hence, we ultimately desire to transform the equations of motion from the inertial frame into the non-inertial frame fixed in the vehicle.

3) If \vec{V}_0 is some vector measure in the x_0, y_0, z_0 frame and \vec{V} some vector measured in the x, y, z frame which is only changed in orientation then:

$$\vec{V} = T(\phi, \psi, \theta) \vec{V}_0 \text{ where } T(\phi, \psi, \theta) = \text{the transform.}$$

where

$$T(\phi, \psi, \theta) = \begin{bmatrix} \cos\psi \cos\theta \\ -\sin\psi \cos\theta + \sin\theta \sin\psi \cos\phi \\ \sin\theta \sin\psi + \cos\theta \cos\psi \sin\phi \\ \cos\psi \sin\theta \\ \cos\theta \cos\psi + \sin\theta \sin\psi \sin\phi \\ -\sin\psi \cos\theta + \cos\theta \sin\psi \sin\phi \\ -\sin\psi \\ \sin\theta \cos\psi \\ \cos\theta \cos\psi \end{bmatrix}$$

4) If \vec{V}_0 and \vec{V} are the same vectors as in (3) above, then:

$$\vec{V}_0 = T^{-1}(\phi, \theta, \psi) \vec{V}$$

Where

$$T^{-1}(\phi, \theta, \psi) =$$

$\cos\theta \cos\psi$	$-\sin\theta \cos\phi + \sin\theta \sin\phi \cos\psi$
$\cos\theta \sin\psi$	$\cos\phi \cos\psi + \sin\phi \sin\theta \sin\psi$
$-\sin\theta$	$\sin\phi \cos\psi$

$\sin\theta \sin\psi + \cos\theta \cos\phi \sin\psi$
$-\sin\theta \cos\psi + \cos\theta \sin\phi \sin\psi$
$\cos\phi \cos\psi$

APPENDIX B

B.1 LIST OF VARIABLES

<u>VARIABLE</u>	<u>MEANING</u>
A	Six by six matrix containing coefficients of UDT, VDT, WDT, PDT, QDT, RDT.
AA	Six by six matrix set equal to matrix A before each call to FUNC. The values of AA are lost in the matrix reduction performed by LEQTIF.
A1,A2	Limits used for selecting the proper propeller thrust.
AI,BI,C!	Set of constants representing the propeller thrust in the X-equation.
B	Ship's buoyancy.
BORATE	Average bowplane rate.
BOWMAX	Maximum ordered bowplane deflection.
COURSE	New course for the ship.
DECRIT	Value of depth error when dive planes are returned to zero deflection.
DEL1	Calculated deflection of the bowplane.
DEL3	Calculated deflection of the stern plane.

DEL4	Calculated deflection of the rudder.
DELB	Actual bowplane deflection at time t.
DELB0	Actual bowplane deflection (units changed for output).
DELGM	Amount the original GM is to be changed.
DELR	Actual rudder deflection at time t.
DELRO	Actual rudder deflection (units changed for output).
DELS	Actual stern plane deflection at time t.
DELS0	Actual stern plane deflection (units changed for output).
DELT	Time increment used in iteration.
DIFF	Difference between present depth and ordered depth.
E	Six by one matrix containing solution to right hand side of equations of motion.
ICNT	Counter used to count the number of iterations
ID	Forty-character alphanumeric heading.
INDEX	If greater than zero, read new submarine coefficients. If less than or equal to zero, run same submarine for new initial

conditions.

IX,IY,IZ,IXY,IXZ,IYZ	Moments of inertia.
KCOEFF	K - equation coefficients.
L	Ship's overall length.
LEQTIF	Matrix reduction subroutine from IMSLIB [3]
M	Ship's mass.
MAXANG	Maximum ordered dive/ascent angle.
MCOEFF	M - equation coefficients.
NCOEFF	N - equation coefficients.
MODIFF	Acceptable error range around ordered depth.
NOPICH	Acceptable error range around zero pitch angle.
ODEPTH	Ordered depth.
OMCRS	Acceptable error range around ordered course.
P,Q,R	Angular velocity about the x, y, and z axes respectively.
PDT,QDT,RDT	Angular acceleration about the x, y, and z axes respectively.
PDTO,QDTO,RDTO	Angular acceleration about the x, y, and z

axes, respectively (with units changed for output).

PHI	Angle of roll.
PHIO	Angle of roll (with units changed for output).
PO,QO,RO	Angular velocity about the x, y, and z axes respectively (with units changed for output).
PSI	Angle of yaw.
PSIO	Angle of yaw (with units changed for output).
RHO	Sea water density.
RRATE	Average rudder rate.
RUDAMT	Maximum ordered rudder deflection.
STERAT	Average stern plane rate.
STERMX	Maximum ordered stern plane deflection.
STOW	Storage array for half of output data.
T	Present time.
T1 . . . T18	Time lag signals.
THETA	Angle of pitch.

THETA0	Angle of pitch (with units changed for output).
TLGB	Time lag for bowplane control system.
TLGR	Time lag for the rudder control system.
TLGS	Time lag for stern plane control system.
U,V,W	Forward, lateral, and vertical velocities respectively.
UDT,VDT,WDT	Forward, lateral, and vertical accelerations respectively (with units changed for output).
U0	Initial forward velocity.
U00,V0,W0	Forward, lateral, and vertical velocities respectively (with units changed for output).
WT	Ship's weight.
X,Y,Z	Coordinate labels of the fixed (x_0, y_0, z_0) coordinate system.
XB,YB,ZB	The x, y, z position of the center of buoyancy.
XCoeff	X - equation coefficients
XDT,YDT,ZDT	Velocities in the fixed coordinate system along the x_0, y_0, z_0 axes respectively.
XB,YB,ZB	The x, y, z position of the center of

gravity.

YCOEFF

Y - equation coefficients

ZCOEFF

Z - equation coefficients

U U U U U

ۛ ۛ ۛ

```

INDEX=0
1 READ(5,16,END=999) INDEX
  IF(INDEX) 2,2,3
3 READ(5,10,END=999) (XCOEFF(1),1=1,16)
  READ(5,10) (YCOEFF(1),1=1,22)
  READ(5,10) (ZCOEFF(1),1=1,22)
  READ(5,10) (KCOEFF(1),1=1,17)
  READ(5,10) (WCOEFF(1),1=1,23)
  READ(5,10) (NCOEFF(1),1=1,22)

```

```

HEAD(5,10) (A1(1),I=1,3)
HEAD(5,10) (B1(1),I=1,3)
HEAD(5,10) (C1(1),I=1,3)
HEAD(5,13) A1,A2

```

C
C
C

```

READ SHIP CHARACTERISTICS --- MASS, LENGTH, ETC.

```

```

HEAD(5,12) XG,YG,ZG
HEAD(5,12) XB,YB,ZB
HEAD(5,12) M,L,RMU
HEAD(5,12) IX,IY,IZ
HEAD(5,12) IXY,IXZ,IYZ
HEAD(5,13) WT,B

```

C
C
C

```

READ CONSTANTS FOR SHIP CONTROL SURFACES

```

```

HEAD(5,13) K1,K2
HEAD(5,11) K5,K6,K7,K8
HEAD(5,13) K9,K10

```

C
C
C

```

READ INITIAL CONDITIONS --- DEPTH, SPEED, ANGLES, ETC.

```

```

2 HEAD(5,14,END=999) (IU(1),I=1,40)

```

```

ICNT=1

```

```

HEAD(5,15) U,V,W,UOT,VOT,WOT
HEAD(5,15) P,Q,R,POT,QOT,ROT
HEAD(5,12) THETA,PHI,PSI
HEAD(5,18) DELS,DELB,DELR,DELGM,Z

```

C
C
C

```

READ SENSITIVITY PARAMETERS FOR COURSE, DEPTH & PITCH

```

```

HEAD(5,11) MODIFF,NOPICH,UNCGRS,DCRIT

```

```

READ COURSE CHANGE INFORMATION

```

C
C
C

```

HEAD(5,18) TLGR,COURSE,KRATE,RUDANT,DELT

```

READ DEPTH CHANGE INFORMATION

```

READ(5,19)BORATE,BORMAX,STERAT,STERMX,MAXANG,DEPTH,TLAB8,TLA9S
10 FORMAT(D10.4)
11 FORMAT(4D10.4)
12 FORMAT(3D10.4)
13 FORMAT(2D10.4)
14 FORMAT(4D41)
15 FORMAT(6D10.4)
16 FORMAT(12)
18 FORMAT(5D10.4)
19 FORMAT(8D10.4)

```

```

15)' ,6X,'(KTS)'.4X,'(KTS/SEC)'.2X,'(KTS/SEC)'.2X,'(KTS/SEC)'.3X,'(D
2EG)'.7X,'(DEG)')

```

C
C
C

```

INITIALIZE STORAGE MATRIX

```

```

DO 46 J=1,11
DO 46 JJ=1,600
STOM(JJ,J)=0.001

```

```

46 CONTINUE
J=1

```

```

IF GM IS BEING CHANGED CALCULATE NEW VERTICAL CENTER OF BUOYANCY

```

C
C
C

```

ZB=ZB-DELGM

```

C
C
C

```

NON-DIMENSIONALIZE MASS AND MOMENTS OF INERTIA

```

```

M=M/(.5*RW0*(L**3))
DENOM=.5*RW0*(L**5)
IX=IX/DENOM
IY=IY/DENOM
IZ=IZ/DENOM
IXY=IXY/DENOM
IYZ=IYZ/DENOM
IXZ=IXZ/DENOM

```

C
C
C

```

CALCULATE COEFFICIENT VALUES FOR THE SIX LINEAR D.E.'S

```

```

A(1,1)=M-XCOEFF(4)
A(1,2)=0.0
A(1,3)=0.0
A(1,4)=0.0
A(1,5)=M*ZG
A(1,6)=-M*YG
A(2,1)=0.0
A(2,2)=M-YCOEFF(6)

```



```

A(2,3)=0.0
A(2,4)=- (M*ZG+YCOEFF(2)*L)
A(2,5)=0.0
A(2,6)=-M*YG-YCOEFF(1)*L
A(3,1)=0.0
A(3,2)=0.0
A(3,3)=-M-ZCOEFF(5)
A(3,4)=-M*YG
A(3,5)=- (M*YG+ZCOEFF(1)*L)
A(3,6)=0.0
A(4,1)=0.0
A(4,2)=- ((M-ZG/(L**2)) + KCOEFF(8)/L)
A(4,3)=-M*YG/(L**2)
A(4,4)=-IX-KCOEFF(1)
A(4,5)=-IXY
A(4,6)=- (IXZ+KCOEFF(2))
A(5,1)=-M*ZG/(L**2)
A(5,2)=0.0
A(5,3)=- ((M*YG/(L**2))+MCOEFF(6)/L)
A(5,4)=-IXY
A(5,5)=-IY-MCOEFF(1)
A(5,6)=-IYZ
A(6,1)=-M*YG/(L**2)
A(6,2)=- ((M*YG/(L**2))-MCOEFF(6)/L)
A(6,3)=0.0
A(6,4)=- (IXZ+MCOEFF(2))
A(6,5)=-IYZ
A(6,6)=-IY-KCOEFF(1)

```

C INITIALIZE TIME, POSITION, AND RATE QUANTITIES

```

T=0.001
T1=1.
T2=1.
T3=1.
T4=1.

```

C
C
C

T5=1.
 T6=1.
 T7=1.
 T8=1.
 T9=1.
 T10=1.
 T11=1.
 T12=1.
 T13=1.
 T14=1.
 T15=1.
 T16=1.
 T17=1.
 T18=1.
 X=0.001
 Y=0.001
 XDT=0.
 YDT=0.
 ZDT=0.
 ADIFF=0.
 ATHTA=0.
 APHI=0.
 CRS2=100.
 CHECK=0.

CONVERT ANGLES IN DEGREES TO RADIANS AND KNOTS TO FT/SEC FOR INTERNAL USE.

C
C
C

UO=0
 UO=UO*1.6889
 U=U*1.6889
 V=V*1.6889
 W=W*1.6889
 UDT=UDT*1.6889
 VDT=VDT*1.6889
 WDT=WDT*1.6889
 PHI=PHI/57.2958

```

THE TA=THE TA/57.2958
PSI=PSI/57.2958
COURSE=COURSE/57.2958
RRATE=RRATE/57.2958
HWDAMT=HWDAMT/57.2958
BURATE=BURATE/57.2958
BOMBAX=BOMBAX/57.2958
STERAT=STERAT/57.2958
STERMX=STERMX/57.2958
MAXANG=MAXANG/57.2958
ONCRS=ONCRS/57.2958
NOPICH=NOPICH/57.2958
IF(T.EQ.0.001) GO TO 31

C
C
C
C
AS EACH ITERATION BEGINS, INITIALIZE 'AA' MATRIX EQUAL TO 'A' MATRIX.
EACH CALLING TO 'FUNC' WILL DESTROY 'AA' MATRIX DURING GAUSSIAN REDUCTION.

30 DO 53 K=1,6
DO 53 JJ=1,6
AA(K,JJ)=A(K,JJ)
53 CONTINUE

C
C
C
CALL 'FUNC' SUBROUTINE AND CALCULATE ACCELERATIONS AT TIME T + DELTA T.

C
C
CALL FUNC

C
C
UPDATE ANGLES AND VELOCITIES

C
C
C
PHI=PHI+(DELTA*P)+((DELTA**2)/2)*PDT
THE TA=THE TA+(DELTA*Q)+((DELTA**2)/2)*QDT
PSI=PSI+(DELTA*R)+((DELTA**2)/2)*RDT
U=U+DELTA*UDT
V=V+DELTA*VDT
W=W+DELTA*WDT
P=P+DELTA*PDT
Q=Q+DELTA*QDT

```

```

R=R+DELTA*ROT
C
C
C
CALCULATE VELOCITY OF SUBMARINE IN FIXED COORDINATE SYSTEM.
XUT= ((U*(DCOS(THETA)*DCOS(PSI))+V*(-USIN(PSI)*DCOS(PHI)+DSIN(P
111)*DSIN(THETA)*DCOS(PSI))+W*(DSIN(PHI)*DSIN(PSI)+DCOS(PHI)*DCOS(P
211)*DSIN(THETA))*DELTA)
YUT= ((U*(DCOS(THETA)*DCOS(PHI))+V*(DCOS(PHI)*DCOS(PSI)+DSIN(PH
311)*USIN(THETA)*DSIN(PSI))+W*(-USIN(PHI)*DCOS(PSI)+DCOS(PHI)*DSIN(T
211)*USIN(THETA)*DSIN(PSI))*DELTA)
ZUT= ((U*(-DSIN(THETA))*DCOS(THETA))+V*(DCOS(THETA))+W*(DCOS(PHI)*
111)*DCOS(THETA))*DELTA)
C
C
C
CALCULATE POSITION OF SUBMARINE (X,Y,Z) AT TIME T + DELTA T.
X=X+DELTA*XUT
Y=Y+DELTA*YUT
Z=Z+DELTA*ZUT
DIFF=Z-DEPTH
ADIFF=DABS(DIFF)
ATHE TA=DABS(THETA)
APSI=DABS(PSI)
APHI=DABS(PHI)
PI=3.14159
CRS1=COURSE-PI
IF(CRS1.LE.0.) CRS2=DABS(COURSE-PSI)
IF(CRS1.GT.0.) CRS2=DABS(PI-CRS1+PSI)
C
C
C
CALL SUBROUTINES TO ADJUST RUDDER AND PLANES AS NECESSARY TO ACHIEVE
ORDERED COURSE AND DEPTH
C
C
C
IF((CRS2.GT.ONCRS).OR.(DELK.NE.0.)) CALL RUDDER
17 IF(ADIFF.GT.NDIFF) CALL DEPTH
IF(ADIFF.GT.DCRIT) CALL STERN
IF((ADIFF.LT.DCRIT).AND.(ATHE TA.GT.NUPTCH)) CALL STERN
C

```

C CONVERT ANGLES IN RADIANS TO DEGREES AND FT/SEC TO KNOTS FOR OUTPUT

C
C
31 PDTO=PUT*57.2958
QUTO=QOT*57.2958
RDTO=ROT*57.2958
DELRD=DELR*57.2958
DELSU=DELS*57.2958
DELRD=DELR*57.2958
PHIU=PHI*57.2958
UOU=U/1.6889
VO=V/1.6889
WO=W/1.6889
PSIU=PSI*57.2958
PHIU=PHI*57.2958
THE TAO=THE TA*57.2958
UUTO=UT/1.6889
VUTO=VT/1.6889
WUTO=WT/1.6889
PO=P*57.2958
QC=Q*57.2958
RO=R*57.2958

C STORE VALUES THAT CANNOT BE WRITTEN NOW IN MATRIX 'STOM' FOR OUTPUT
C
C LATER WITH A NEW HEADING.

C
C
C
C
STOM(J,1)=T
STOM(J,2)=PHIO
STOM(J,3)=THE TAO
STOM(J,4)=PSIO
STOM(J,5)=PU
STOM(J,6)=QO
STOM(J,7)=RO
STOM(J,8)=PDTO
STOM(J,9)=QUTO
STOM(J,10)=RDTO
STOM(J,11)=DELRD

```

WRITE(6,50)T,X,Y,Z,U00,V0,W0,U0T0,V0T0,W0T0,DEL50,DELBO
50 FORMAT(3(1X,D10.3),D11.4,8(1X,D10.3),/)

C
C
C
TIME INCREMENT ALGORITHM.

21 T=T+DELT
J=J+1

C
C
END CALCULATIONS WHEN ORDERED COURSE AND DEPTH IS REACHED
IF((CRS2.LE.ONCRS).AND.(ADIFF.LT.MODIFF).AND.(ATHETA.LE.MOPICH)).AN
10.(APHI.LE.1.0)) GO TO 39

C
C
END CALCULATIONS WHEN 'ICNT' NUMBER OF ITERATIONS ARE COMPLETE.

IF (ICNT.GE.600) GO TO 39
ICNT=ICNT+1
GO TO 30

C
C
C
WRITE NEW HEADING FOR OUTPUT THAT HAS BEEN STORED IN MATRIX 'STOW'.

39 WRITE(6,48) (ID(1),I=1,40)
48 FORMAT(1H1,39X,'.....')
1.'./,40X,'.',49X,'.',40X,'.',40A1,'.',40X,'.',49X,'.',
2./,40X,'.....',/)
WRITE(6,44)
44 FORMAT(5X,'T',11X,'PHI',6X,'THETA',6X,'PSI',10X,'P',10X,'Q',10X,'R
.',9X,'PUT',8X,'QUT',8X,'ROT',7X,'DELR')
WRITE(6,47)
47 FORMAT(3X,'(SEC)',6X,'(DEG)',6X,'(DEG)',6X,'(DEG)',4X,'(DEG/SEC)',
1,2X,'(DEG/SEC)',2X,'(DEG/SEC) (DEG/SEC2) (DEG/SEC2) (DEG/SEC2)',4X
2,'(DEG)')
WRITE(6,43)((STOW(J,I),I=1,11),J=1,ICNT)
43 FORMAT(11(1X,D10.3),/)

C
C
CHECK TO SEE IF NEW DATA IS WAITING FOR ANOTHER RUN.

```

C
GO TO 1
949 STOP
LMU

SUBROUTINE FUNC

```

      IMPLICIT REAL*8(A-H),REAL*8(L-Z)
      REAL*8 XCOEFF(16),YCOEFF(22),ZCOEFF(22),KCOEFF(17),MCOEFF(23),NCOE
      IFF(22),AA(6,6),E(6),AI(3),BI(3),CI(3),WKAREA(6,6)
      REAL*8 IX,IY,IZ,IXY,IXZ,IYZ
      COMMON /FOUR/ U,V,W,UO,PHI,AI,BI,CI,XCOEFF,YCOEFF,ZCOEFF,KCOEFF,MCOEFF,NCOEFF,L,M,P,
      WKU,WT,B,XG,YG,ZG,IX,IY,IXZ,IXY,IYZ,RDT,QDT,P
      IODEFF,MCOEFF,L,M,P,WKU,WT,B,XG,YG,ZG,IX,IY,IXZ,IXY,IYZ,RDT,QDT,P
      ZDT,WDT,QDT,XB,YB,ZB,AI,AZ,AA,WKAREA
      COMMON /FIVE/ THETA,Q,DELS,DELB,DELT,PSI,R,DELR

```

```

      SUBROUTINE FUNC CALCULATES THE RIGHT HAND SIDE OF THE SIX NON-LINEAR
      DIFFERENTIAL EQUATIONS.

```

```

      ETA=0.001

```

```

      CALCULATE MAGNITUDE OF VECTOR VELOCITY

```

```

      BIGU=DSQRT(U**2+V**2+W**2)
      ETA=UO/BIGU
      THETA=THETA
      PHI=PHI
      VM=DSQRT(V**2+W**2)

```

```

      TEST FOR APPROPRIATE THRUST/DRAG COEFFICIENT

```

```

      IF(ETA.LE.A1) GO TO 10
      IF((ETA.GT.A1).AND.(ETA.LE.A2)) GO TO 11
      IF(ETA.GT.A2) GO TO 12
10  AI=AI(1)
      BI=BI(1)
      CI=CI(1)
      GO TO 40
11  AI=AI(2)
      BI=BI(2)

```


C11-C1(2)
G0 TO 40
12 A11-A1(3)
B11-B1(3)
C11-C1(3)

RIGHT-HAND-SIDE OF LINEAR D.E.'S FOR QDT, VDT, WDT, PDT, QOT, AND RDT.

40 E(1) = (L * (XCOEFF(1) * (Q**2) + XCOEFF(2) * (R**2) + XCOEFF(3) * R * P)) + (XCOEFF(1(5) * V * R + XCOEFF(6) * W * Q) + ((1./L) * (XCOEFF(7) * (U**2) + XCOEFF(8) * (V**2) + XCOEFF(9) * (W**2) + A11 * (U**2) + B11 * U * W + C11 * (U**2)) + ((1./L) * (U**2) * XCOEFF(10) * (DEL R**2) + XCOEFF(11) * (DELS**2) + XCOEFF(12) * (DELB**2)) + XCOEFF(13) * (ETA-1.) * (XCOEFF(13) * (V**2) + XCOEFF(14) * (W**2) + XCOEFF(15) * (DELR * U)**2) + XCOEFF(16) * ((DELS * U)**2)) - ((2./((W * U) * (L**3))) * (WT-8) * 6DSIN(THETAR)) - (H * (W * Q - V * R - AU * ((Q**2) + (R**2) + V * G * P = Q + Z * G * P * R))

VV=V

IF (VV.EQ.0.001) VV=0.10-15

E(2) = L * (YCOEFF(3) * P * DABS(P) + YCOEFF(4) * P * Q + YCOEFF(5) * Q * R) + (YCOEFF(7(1) * Q * V + YCOEFF(8) * W * P + YCOEFF(9) * W * R + YCOEFF(10) * U * R + YCOEFF(11) * U * P + YC2OEFF(12) * U * DABS(R) * DELR + YCOEFF(13) * (V/DABS(VW)) * DABS(VW) * DABS(R) + Y3COEFF(19) * U * R * (ETA-1)) + ((1./L) * (YCOEFF(14) * (U**2) + YCOEFF(15) * U * V + YC4OEFF(16) * V * DABS(VW) + YCOEFF(17) * V * W + YCOEFF(18) * (U**2) * DELR)) + ((2./R5HW) * L**3)) * (WT-8) * DCOS(THETAR) * DCOS(PHIR)) + (((1./L) * (ETA-1)) * (YCOEFF(20) * U * V + YCOEFF(21) * V * DABS(VW) + YCOEFF(22) * DELR * (U**2))) - (H * (U * R - W * 7P - YG * (R**2 + P**2) * Z * G * Q * R + XG * Q * P))

WW=W

IF (WW.EQ.0.001) WW=0.10-15

E(3) = (L * (ZCOEFF(2) * P**2 + ZCOEFF(3) * (W**2) + ZCOEFF(4) * R * P)) + ZCOEFF(6(1) * V * R + ZCOEFF(7) * V * P + ZCOEFF(8) * U * Q + ZCOEFF(9) * U * DABS(Q) * DELS + ZCOEFF(120) * (W/DABS(WW)) * DABS(VW) * DABS(Q) + ZCOEFF(11) * U * Q * (ETA-1)) + ((1./L) * (ZC3OEFF(12) * (U**2) + ZCOEFF(13) * U * W + ZCOEFF(14) * W * DABS(VW) + ZCOEFF(15) * U * 4DABS(W) + ZCOEFF(16) * DABS(W * W) + ZCOEFF(17) * (V**2) + ZCOEFF(18) * (U**2) *

```

5UELS+ZCOEFF(19)*(U**2)*DELBI)*((1/L)*(ETA-1))*(ZCOEFF(20)*U**W+ZCO
6EFF(21)*W*DABS(VW)+ZCOEFF(22)*DELS*(U**2))+((2/(WU)*L**3))*(WT-B)*
7UCOS(THETAR)*DCOS(PHIR))- (M*(V*P-U*Q-ZG*(P**2)+(Q**2))+XG*R*P+YG*
8R*Q))

```

C

```

E(4)=KCOEFF(3)*Q*R+KCOEFF(4)*P*Q+KCOEFF(5)*P*DABS(P)+((1/L)*(KCOEFF
1F(6)*U*P+KCOEFF(7)*U*R+KCOEFF(9)*V*Q+KCOEFF(10)*W*P+KCOEFF(11)*W*R
2))+((1/L**2)*(KCOEFF(12)*(U**2)+KCOEFF(13)*V*U+KCOEFF(14)*V*DABS(V
3W)+KCOEFF(15)*V*W+KCOEFF(16)*(U**2)*DEL R+KCOEFF(17)*(U**2)*(ETA-1)
4))*((2/(WU)*L**5))*((YG*WT-YB*B)*DCOS(THETAR)*DCOS(PHIR)-(ZG*WT-ZB
5*B)*DCOS(THETAR)*DSIN(PHIR)))-((1Z-1Y)*Q*R)+(1XZ*Q*P)-((R**2-Q**2)
6*1YZ)-1XV*P*R-((M*(1/L**2))*(YG*(V*P-U*Q)-ZG*(U*R-W*P)))

```

C

```

E(5)=(MCOEFF(2)*P**2+MCOEFF(3)*(R**2)+MCOEFF(4)*R*P+MCOEFF(5)*Q*DA
13S(Q))+((1/L)*(MCOEFF(7)*V*R+MCOEFF(8)*V*Q+MCOEFF(9)*U*Q+MCOEFF(10
2)*U*DABS(Q)*DELS+MCOEFF(11)*DABS(VW)*Q+MCOEFF(20)*U*Q*(ETA-1))+((
31/L**2)*(MCOEFF(12)*(U**2)+MCOEFF(13)*U*W+MCOEFF(14)*W*DABS(VW)+MC
4OUFF(15)*U*DABS(W)+MCOEFF(16)*DABS(W*W)+MCOEFF(17)*(V**2)+MCOEFF(
518)*(U**2)*DELS+MCOEFF(19)*(U**2)*DELB))+((2/(WU)*L**5))*((XG*WT-X
6B*B)*DCOS(THETAR)*DCOS(PHIR)-(ZG*WT-ZB*B)*DSIN(THETAR))+((1/L**2)
7*(ETA-1)*(MCOEFF(21)*U*W+MCOEFF(22)*W*DABS(VW)+MCOEFF(23)*DELS*(1*
8*2))-((1Y-1X)*P*Q+1YZ*R*P-1XV*(Q**2)-(P**2))-1XZ*R*Q-(M*(1/L**2)*
9(XG*(U*R-W*P)-YG*(W*Q-V*R)))

```

C

```

E(6)=MCOEFF(3)*P*Q+MCOEFF(4)*Q*R+MCOEFF(5)*R*DABS(R)+((1/L)*(MCOEFF
1(7)*W*R+MCOEFF(8)*W*P+MCOEFF(9)*V*Q+MCOEFF(10)*U*P+MCOEFF(11)*U*R+
2MCOEFF(12)*U*DABS(W)*DEL R+MCOEFF(13)*DABS(VW)*R)+((1/L**2)*(MCOEFF
3F(14)*U**2+MCOEFF(15)*U*V+MCOEFF(16)*V*DABS(VW)+MCOEFF(17)*W*V+MCO
4LEFF(18)*(U**2)*DEL R))+((2/(WU)*L**5))*((XG*WT-XB*B)*DCOS(THETAR)*
5DSIN(PHIR)+(YG*WT-YB*B)*DSIN(THETAR))+((1/L)*(ETA-1)*MCOEFF(19)
6*U*R)+((1/L**2)*(ETA-1)*(MCOEFF(20)*U*V+MCOEFF(21)*V*DABS(VW)+MC
7UEFF(22)*DEL R*U**2))-((1Y-1X)*P*Q)+(1YZ*P*R)-(1XV*(Q**2-P**2))-((1X
8Z*P*R)-(M*(1/L**2)*(XG*(U*R-W*P)-YG*(W*Q-V*R)))

```

C

THIS CALL PERFORMS THE GAUSSIAN REDUCTION OF THE 'AA' MATRIX.

C

LEQTIF IS A LIBRARY FUNCTION IN SYSS.INSLIB.

C

C
CALL LEQTF (AA,1,6,6,E,3,WKARLA,IER)
RDT=E(6)
QOT=E(5)
PUT=E(4)
WUT=E(3)
VUT=E(2)
UUT=E(1)
RETURN
END

[illegible]

IF(T6.LE.TLAGR) GO TO 3
 IF(DABS(DEL4).LT.DELR) GO TO 5
 DELR=DEL4+DELTA

C
 C
 C

ENSURE MAX. ORDERED RUDDER DEFLECTION IS NOT EXCEEDED.

14 IF(DEL4.GE.RUDANT) DELR=RUDANT
 GO TO 99

C
 C
 C

ENSURE TIME LAG HAS PASSED BEFORE RUDDER MOVES.

5 T11=0.
 IF(T12.LE.TLAGR) GO TO 7

C
 C
 C

AS ERROR SIGNAL GETS SMALLER, RUDDER ANGLE CAN BE REDUCED.

DEL4=DEL4-DABS(DEL4)
 IF(DEL4.LT.DELTA) DELR=DEL4-DELTA
 IF(DEL4.GT.DELTA) DELR=DEL4+DELTA
 GO TO 99

C
 C
 C

ENSURE TIME LAG HAS PASSED BEFORE RUDDER MOVES.

1 T6=0.
 IF(T5.LE.TLAGR) GO TO 4
 IF(DABS(DEL4).LT.DABS(DELR)) GO TO 6
 DELR=DEL4-DELTA

C
 C
 C

ENSURE MAX. ORDERED RUDDER DEFLECTION IS NOT EXCEEDED.

10 IF(DABS(DEL4).GE.RUDANT) DELR=RUDANT
 GO TO 99

C
 C
 C

ENSURE TIME LAG HAS PASSED BEFORE RUDDER MOVES.

6 T12=0.

```

IF(T11.LE.TLAGR) GO TO 8
C
C
C
AS ERROR SIGNAL GETS SMALLER, RUDDER ANGLE CAN BE REDUCED.
DELTA=UABS(DELTA)-UABS(DELTA)
IF(DELTA.LT.DELTA) DELTA=DELTA+DELTA
IF(DELTA.GT.DELTA) DELTA=DELTA+DELTA
GO TO 99
C
C
C
ALGORITHM FOR SMALL ERROR SIGNALS.
18 IF(R.LT.O.) GO TO 11
IF(R.GT.O.) GO TO 12
GO TO 99
C
C
C
ENSURE TIME LAG HAS PASSED BEFORE RUDDER MOVES.
11 T17=0.
IF(T18.LE.TLAGR) GO TO 9
DELTA=DELTA-DELTA
IF((DELTA.LT.O.).AND.(DCORR.LE.O.)) GO TO 15
GO TO 10
C
C
C
ENSURE TIME LAG HAS PASSED BEFORE RUDDER MOVES.
12 T18=0.
IF(T17.LE.TLAGR) GO TO 13
DELTA=DELTA+DELTA
IF((DELTA.GT.O.).AND.(DCORR.GE.O.)) GO TO 15
GO TO 14
15 DELTA=0.
GO TO 99
9 T18=T18+DELTA
GO TO 99
13 T17=T17+DELTA
GO TO 99

```

3 16-16+DEL I
GO 10 99
4 15-15+DEL I
GO 10 99
7 112-112+DEL I
GO 10 99
8 111-111+DEL I
99 RETURN
END

```

SUBROUTINE DEPTH
C
IMPLICIT REAL*8(A-H), REAL*8(L-Z)
REAL*8 K1,K2,K3,K4,K5,K6,K7,K8,K9,K10
COMMON /TWO/ K1,K2,T1,T2,T13,T14,T15,T16,TLAGB,BURATE,BOMMAX,T,CHE
ICK
COMMON /FIVE/ THETA,Q,DELS,DELB,DELT,PSI,R,DELR
COMMON /SIX/ DIFF,ADIFF,ZDT,ATHETA,MAXANG,NOPICH,DCRIT

CALCULATED DIVE PLANE ANGLE IS PROPORTIONAL TO DEPTH ERROR
SIGNAL AND RATE OF CHANGE OF DEPTH.
C
DELT=K1*DIFF+K2*ZDT
ADELT=DABS(DELT)
DELTA=BURATE*DELT
DELTB=DABS(DELB)-ADELT
C
IF THE ERROR SIGNAL IS SMALL GO TO 18 FOR FINER CONTROL.
C
C
IF(ADIFF.LT.DCRIT) GO TO 18
IF(DELT.LT.O.) GO TO 4
C
ENSURE TIME LAG HAS PASSED BEFORE PLANE MOVES.
C
C
T2=0.
IF(T1.LE.TLAGB) GO TO 89
C
IF MAX DIVE/ASCENT ANGLE IS EXCEEDED TAKE CORRECTIVE ACTION.
C
C
5 IF((ATHETA.GE.MAXANG).AND.(THETA.GE.O.)) GO TO 2
DELB=DELB+DELTA
IF(DELB.GT.ADELT) GO TO 10
C
ENSURE MAX PLANE ANGLE IS NOT EXCEEDED.
C
C

```



```

9 IF (DEL B .GE. BOMMAX) DELB=BOMMAX
C
C
C
  IF MAX DIVE/ASCENT ANGLE IS EXCEEDED WRITE DIAGNOSTIC.
C
C
  IF (ATHETA .GE. MAXANG) GO TO 6
  GO TO 91
2 DELB=DELB-DELTA
  GO TO 8
C
  ENSURE TIME LAG HAS PASSED BEFORE PLANE MOVES.
C
C
C
10 T15=0.
  IF (T16 .LE. TLAGB) GO TO 88
C
C
C
  AS ERROR SIGNAL GROWS SMALLER, THE PLANE ANGLE CAN BE REDUCED.
C
C
C
  IF (DELT B .LE. DELTA) DELB=ADELT
  IF (DELT B .GT. DELTA) DELB=DELB-DELTA
  GO TO 92
C
  ENSURE TIME LAG HAS PASSED BEFORE PLANE MOVES.
C
C
C
4 T1=0.
  IF (T2 .LE. TLAGB) GO TO 90
C
C
C
  IF MAX DIVE/ASCENT ANGLE IS EXCEEDED TAKE CORRECTIVE ACTION.
C
C
C
  IF ((ATHETA .GE. MAXANG) .AND. (THETA .LE. 0.)) GO TO 3
  DELB=DELB-DELTA
  IF (DABS(DELB) .GT. ADEL1) GO TO 11
C
  ENSURE MAX PLANE ANGLE IS NOT EXCEEDED.
C
C
C
8 IF (DABS(DELB) .GE. BOMMAX) DELB=-BOMMAX
C
C
C
  IF MAX DIVE/ASCENT ANGLE IS EXCEEDED WRITE DIAGNOSTIC.
C
C
C

```

```

IF(ATMETHA.GE.MAXANG) GO TO 6
GO TO 91
3 DELB=DELB+DELTA
GO TO 9

```

```

C
C ENSURE TIME LAG HAS PASSED BEFORE PLANE MOVES.
C

```

```

11 T16=0.
IF(T15.LE.TLAGB) GO TO 87

```

```

C AS ERROR SIGNAL GROUPS SMALLER, THE PLANE ANGLE CAN BE REDUCED.
C

```

```

IF(DELTA.LE.DELTA) DELB=-ADEL1
IF(DELTA.GT.DELTA) DELB=DELTA+DELTA
GO TO 92

```

```

C DIAGNOSTIC WRITE STATEMENT
C

```

```

6 WRITE(6,7)T
7 FORMAT(/,20X,'***** EXCEEDED MAX DIVE/ASCENT ANGLE AT TIME =',F10
  4,' ',/20X,'***** STANDARD FIXUP TAKEN *****')
GO TO 91

```

```

C CONTROL FOR SMALL ERROR SIGNALS.
C

```

```

18 IF((ZDT.LT.0.).AND.(DIFF.GT.0.).AND.(DELB.EQ.0.)) RETURN
IF((ZDT.GT.0.).AND.(DIFF.LT.0.).AND.(DELB.EQ.0.)) RETURN
IF((ZDT.GT.0.).AND.(DIFF.GT.0.).AND.(DELB.LE.4.)) GO TO 14
IF((ZDT.GT.0.).AND.(DIFF.LT.0.).AND.(DELB.NE.0.)) GO TO 12
IF((ZDT.GT.0.).AND.(DIFF.GT.0.).AND.(DELB.GT.4.)) RETURN
IF((ZDT.LT.0.).AND.(DIFF.GT.0.).AND.(DELB.NE.0.)) GO TO 12
IF((ZDT.LT.0.).AND.(DIFF.LT.0.).AND.(DELB.GE.-4.)) GO TO 13
IF((ZDT.LT.0.).AND.(DIFF.LT.0.).AND.(DELB.LT.-4.)) RETURN

```

```

12 IF(DELB.LT.0.) GO TO 22
CHECK-2.

```

ENSURE TIME LAG HAS PASSED BEFORE PLANE MOVES.

```

13 T13=0.
  IF (T14.LE.TLAGB) GO TO 17
  IF (CHECK.EQ.2.) GO TO 23
  IF (DELB.GT.-6.) GO TO 20
  GO TO 91
23 IF ((DELB-DELTA) .LT.0.) GO TO 15
20 DELB=DELB-DELTA
  GO TO 92
22 CHECK=1.

```

ENSURE TIME LAG HAS PASSED BEFORE PLANE MOVES.

```

14 T14=0.
  IF (T13.LE.TLAGB) GO TO 16
  IF (CHECK.EQ.1.) GO TO 21
  IF (DELB.LT.6.) GO TO 19
  GO TO 91
21 IF ((DELB+DELTA).GT.0.) GO TO 15
19 DELB=DELB+DELTA
  GO TO 92
15 DELB=0.
  GO TO 91
16 T13=T13+DELT
  GO TO 91
17 T14=T14+DELT
  GO TO 91
18 T15=T15+DELT
  GO TO 91
19 T16=T16+DELT
  GO TO 91
20 T1=T1+DELT
  GO TO 91
21 T2=T2+DELT

```

GO TO 91
92 IF (DELB.LI.-BOMMAX) DELB--BOMMAX
IF (DELB.GI.-BOMMAX) DELB-BOMMAX
91 RETURN
END

```

SUBROUTINE STERN
C
  IMPLICIT REAL*8(A-H), REAL*8(L-Z)
  REAL*8 K1,K2,K3,K4,K5,K6,K7,K8,K9,K10
  COMMON /ONE/ K3,K4,K5,K6,K7,K8,K9,K10,ILAGS,STERAT,STE
  1000
  COMMON /FIVE/ THE TA,U,DELS,DELB,DELT,PSI,R,DELR
  COMMON /SIX/ UIFF,ADIFF,ZDT,ATHETA,MAXANG,NOPICH,DCRIT
C
  CALCULATED PLANE ANGLE IS PROPORTIONAL TO DEPTH ERROR, RATE OF
  CHANGE OF DEPTH, PITCH ANGLE, AND RATE OF CHANGE OF PITCH.
C
  DEL3=K5*DIFF+K6*ZDT+K7*THE TA+K8*U
  ADL3=DABS(DEL3)
  DELTA=STERAT*DELT
  DELTB=DABS(DELS)-ADL3
C
  FOR SMALL DEPTH ERROR AND LARGE PITCH ERROR GO TO 18
C
  IF((ADIFF.LT.DCRIT).AND.(ATHETA.GT.NOPICH)) GO TO 18
  IF(DEL3.GT.0.) GO TO 2
C
  ENSURE TIME LAG HAS PASSED BEFORE PLANE MOVES.
C
  T3=0.
  IF(T4.LE.ILAGS) GO TO 8
C
  IF MAX DIVE/ASCENT ANGLE IS EXCEEDED, TAKE CORRECTIVE ACTION.
C
  IF((ATHETA.GE.MAXANG).AND.(THE TA.LE.0.)) GO TO 5
  IF(DELS.GT.ADEL3) GO TO 6
  DELS=DELS+DELTA
C
  ENSURE MAX ORDERED PLANE ANGLE IS NOT EXCEEDED.
C

```

```

1 IF(DELS.GE.STERMAX) DELS=STERMAX
  GO TO 99
  C
  C
  C
  ENSURE TIME LAG HAS PASSED BEFORE PLANE MOVES.
  C
  C
  C
2 T4=0.
  IF(TJ.LE.TLAGS) GO TO 9
  C
  C
  C
  IF MAX DIVE/ASCENT ANGLE IS EXCEEDED, TAKE CORRECTIVE ACTION.
  C
  C
  C
  IF((ATHETA.GE.MAXANG).AND.(THETA.GE.O.)) GO TO 4
  IF(DABS(DELS)-GT.ADEL3) GO TO 7
  DELS=DELS-DELTA
  C
  C
  C
  ENSURE MAX ORDERED PLANE ANGLE IS NOT EXCEEDED.
  C
  C
  C
3 IF(DABS(DELS).GE.STERMAX) DELS=-STERMAX
  GO TO 99
  C
  C
  C
4 DELS=DELS+DELTA
  GO TO 1
  C
  C
  C
5 DELS=DELS-DELTA
  GO TO 3
  C
  C
  C
  ENSURE TIME LAG HAS PASSED BEFORE PLANE MOVES.
  C
  C
  C
6 T8=0.
  IF(T7.LE.TLAGS) GO TO 10
  C
  C
  C
  AS ERROR SIGNAL GROWS SMALLER, THE PLANE ANGLE CAN BE REDUCED.
  C
  C
  C
  IF(DELTB.LE.DELTA) DELS=ADEL3
  IF(DELTB.GT.DELTA) DELS=DELS-DELTA
  GO TO 99
  C
  C
  C
  ENSURE TIME LAG HAS PASSED BEFORE PLANE MOVES.
  C
  C
  C

```

```

7 T9=0.
  IF(T8.LE.TLAGS) GO TO 11
  AS ERROR SIGNAL GROWS SMALLER, THE PLANE ANGLE CAN BE REDUCED.
  IF(DEL.TB.LE.DELTA) DELS=-ADEL3
  IF(DEL.TB.GT.DELTA) DELS=DELS+DELTA
  GO TO 99
8 T4=T4+DELT
  GO TO 99
9 T3=T3+DELT
  GO TO 99
10 T7=T7+DELT
  GO TO 99
11 T8=T8+DELT
  GO TO 99
12 IF(DELS.LT.0.) GO TO 14
  ENSURE TIME LAG HAS PASSED BEFORE PLANE MOVES.
13 T9=0.
  IF(T10.LE.TLAGS) GO TO 17
  IF((DELS-DELTA).LT.0.) GO TO 15
  DELS=DELS-DELTA
  GO TO 99
  ENSURE TIME LAG HAS PASSED BEFORE PLANE MOVES.
14 T10=0.
  IF(T9.LE.TLAGS) GO TO 16
  IF((DELS+DELTA).GT.0.) GO TO 15
  DELS=DELS+DELTA
  GO TO 99
15 DELS=0.
  GO TO 99
16 T9=T9+DELT

```

```

GO TO 99
17 TIO=TIO+DELT
GO TO 99

C
C
C      CONTROL TO CORRECT PITCH ANGLE.

18 IF ((THE TA.LT.O.) .AND. (Q.GT.O.) .AND. (DELS.EQ.O.)) RETURN
   IF ((THE TA.GT.O.) .AND. (Q.LT.O.) .AND. (DELS.EQ.O.)) RETURN
   IF ((THE TA.LT.O.) .AND. (Q.LT.O.) .AND. (DELS.LT.O.)) RETURN
   IF ((THE TA.GT.O.) .AND. (Q.GT.O.) .AND. (DELS.GT.O.)) RETURN
   IF ((THE TA.LT.O.) .AND. (Q.GT.O.) .AND. (DELS.NE.O.)) GO TO 12
   IF ((THE TA.GT.O.) .AND. (Q.LT.O.) .AND. (DELS.NE.O.)) GO TO 12
   IF ((THE TA.LT.O.) .AND. (Q.LT.O.) .AND. (DELS.GT.O.)) GO TO 13
   IF ((THE TA.GT.O.) .AND. (Q.GT.O.) .AND. (DELS.LT.O.)) GO TO 14
   IF ((THE TA.GT.O.) .AND. (Q.GT.O.) .AND. (DELS.EQ.O.)) GO TO 19
   IF ((THE TA.LT.O.) .AND. (Q.LT.O.) .AND. (DELS.EQ.O.)) GO TO 20

99 RETURN
END

```


B.3 LEQTIF

A - input matrix of dimension N by N containing the coefficient matrix of the equation $AX = B$.

On output, A is replaced by the LU decomposition of a rowwise permutation of A.

M - number of right-hand sides. (input)

N - order of A and number of rows in B. (input)

IA - number of rows in the dimension statement for A and B in the calling program. (input)

B - input matrix of dimension N by M containing right-hand sides of the equation $AX = B$.

On output, the N by M solution X replaces B.

IDGT - input option.

If IDGT is greater than 0 the elements of A and B are assumed to be correct to IDGT Decimal digits and the routine performs an accuracy test.

If IDGT equals zero, the accuracy test is bypassed.

WKAREA - work area of dimension greater than or equal to N .

IER - error parameter

terminal error = $128 + N$

$N = 1$ indicates that A is algorithmically singular.

Warning error = $32 + N$.

$N = 2$ indicates that the accuracy test failed. The computed solution may be in error by more than can be accounted for by the uncertainty of the data.

CALL LEQTIF (A, M, N, IA, B, IDGT, WKAREA, IER)